

# Cost Modeling for Space Optical Telescope Assemblies

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## ABSTRACT

Parametric cost models are used to plan missions, compare concepts and justify technology investments. This paper reviews an on-going effort to develop cost modes for space telescopes. This paper summarizes the methodology used to develop cost models and documents how changes to the database have changed previously published preliminary cost models. While the cost models are evolving, the previously published findings remain valid: it costs less per square meter of collecting aperture to build a large telescope than a small telescope; technology development as a function of time reduces cost; and lower areal density telescopes cost more than more massive telescopes.

**Keywords:** Space Telescope Cost Model, Parametric Cost Model

## 1. INTRODUCTION

Parametric cost models for space telescopes provide several benefits to designers and space system project managers. They identify major architectural cost drivers and allow high-level design trades. They enable cost-benefit analysis for technology development investment. And, they provide a basis for estimating total project cost. A survey of historical models found that there was no definitive space telescope cost model [1]. Thus, there is a need for parametric space telescopes cost models. An effort is underway to develop single variable [2] and multi-variable [3] parametric space telescope cost models based on the latest available data and applying rigorous analytical techniques. Since the publication of the single and multi-variable parametric models, the data base has changed. New telescopes were added to the data base; data for other telescopes was revised; and, other telescopes were removed from the modeling analysis. As a result of these changes, the cost models have changed. But the general findings remain unchanged: aperture diameter is the primary cost driver for large space telescopes; technology development as a function of time reduces cost; it costs less per square meter of collecting aperture to build a large telescope than a small telescope; and it costs more per kg to build a low areal density telescope than a massive telescope.

## 2. METHODOLOGY

Cost and engineering data has been collected on 59 different parameters for 39 x-ray, UV, optical, infrared, microwave and radio space telescopes. But to date, only the 32 normal-incidence UV, Optical, Infrared (UVOIR) missions have been studied for cost modeling. And, of these 32, sufficient data exists for only 15 with which to develop an Optical Telescope Assembly (OTA) cost model (Table 1). Data was collected from multiple sources, including: NAFCOM (NASA/ Air Force Cost Model) database, RSIC (Redstone Scientific Information Center), REDSTAR (Resource Data Storage and Retrieval System), project websites, and interviews.

Table 1: UV/OIR Cost Model Missions Database	
UV/Optical Telescopes	Infrared Telescopes
Commercial #1	Herschel
Commercial #2	IRAS
GALEX	JWST
HiRISE	Planck
HST	SOFIA
IUE	Spitzer
Kepler	WIRE
	WISE

For our study, Optical Telescope Assembly (OTA) is defined as the space observatory subsystem which collects electromagnetic radiation and focuses it (focal) or concentrates it (afocal). An OTA consists of the primary mirror, secondary mirror, auxiliary optics and support structure (such as optical bench or truss structure, primary support structure, secondary support structure or spiders, etc.). An OTA does not include science instruments or spacecraft subsystems. And, cost is defined as prime contract cost without any NASA labor or overhead. Total mission cost is defined as Phase A-D cost, excluding: launch cost; costs associated with NASA labor (civil servant or support contractors) for program management, technical insight/oversight; or any NASA provided ground support equipment, e.g. test facilities. Accounting for NASA overheads would increase the cost by at least 10% and maybe as much as 33%.

Statistical correlations have been evaluated between 18 of 59 variables. And, these parameters have been used to develop single variable cost estimating relationships (CERs) which are evaluated for their ‘goodness’. The variables are divided between technical (Aperture Diameter, PM Focal Length, System Focal Length, Field of View, Pointing Stability, OTA Mass, Total Mass, Spectral Range Minimum, Wavelength of Diffraction Limit, Operating Temperature, Average Input Power, Data Rate, Design Life, and Orbit) and programmatic (Technology Readiness Level, Year of Development, Development Period, and Launch Year). Additionally, insight is gained by combining independent variables to form collector variables (F/#, Volume, Areal Cost or Cost Density).

Two single variable cost estimating relationships (CERs) are reported in this paper. These CERs estimate OTA cost as a function of OTA diameter and OTA mass.

### 3. MODEL CREATION

The first step in creating a statistical cost model is to start with the Cross Correlation Matrix (Figure 1) and look for variables which are highly correlated with cost. When using a cross-correlation matrix, there are several things to consider. First, the higher the correlation value, the greater the cost variation explained by that variable. Second, the sign of correlation is important. It must be consistent with known engineering design principals and manufacturing processes. Third, for multi-variable models, we want variables which independently effect cost. Variables which ‘cross-talk’ with each other are multicollinear.

rev. 5.3.10	Total Cost	OTA Cost	Aperture Diameter	PM F Len.	System F Len.	FOV	Pointing Stability (Arc-Sec / Sec)	Total Mass (kg)	OTA Mass (kg)	Spectral Range minimum (Å)	Diff Lim. λ (Å)	Operating Temp (K)	Avg Input Power (Watts)	Data Rate (Kbps)	Design Life (months)	TRL	Year of Dev. (year)	Dev Period (months)	Date of Launch (year)	Orbit
units	(FY11SM)	(FY11SM)	(m)	(E)	(E)	(°)														
Total Cost	<b>1.00</b>	<b>0.90</b>	<b>0.87</b>	<b>0.84</b>	<b>0.84</b>	-0.12	<b>-0.67</b>	<b>0.96</b>	<b>0.80</b>	0.00	-0.11	0.15	<b>0.72</b>	0.02	<b>0.72</b>	-0.40	0.05	<b>0.78</b>	0.22	0.42
OTA Cost		<b>1.00</b>	<b>0.85</b>	<b>0.91</b>	<b>0.80</b>	-0.12	<b>-0.82</b>	<b>0.91</b>	<b>0.92</b>	-0.21	-0.25	0.23	0.52	-0.13	<b>0.78</b>	-0.50	0.01	<b>0.55</b>	0.22	0.05
Aperture Diameter			<b>1.00</b>	<b>0.87</b>	<b>0.86</b>	-0.17	<b>-0.89</b>	<b>0.90</b>	<b>0.89</b>	0.09	0.16	0.24	0.56	-0.10	<b>0.68</b>	-0.36	0.23	0.60	0.42	0.16
PM F Len.				<b>1.00</b>	<b>0.76</b>	-0.10	<b>-0.81</b>	<b>0.82</b>	<b>0.88</b>	-0.37	-0.22	0.34	0.49	-0.08	<b>0.79</b>	-0.41	0.12	<b>0.66</b>	0.32	0.04
System F Len.					<b>1.00</b>	<b>-0.62</b>	-0.36	<b>0.83</b>	<b>0.74</b>	-0.21	-0.23	0.31	0.41	-0.28	<b>0.82</b>	-0.43	-0.04	0.33	0.12	0.26
FOV						<b>1.00</b>	-0.32	-0.10	-0.21	0.29	<b>0.38</b>	-0.04	0.05	0.21	-0.27	0.36	0.36	0.13	0.35	0.16
Pointing Stability							<b>1.00</b>	<b>-0.71</b>	<b>-0.87</b>	0.31	0.33	-0.16	<b>-0.66</b>	0.20	-0.58	0.23	0.09	<b>-0.76</b>	-0.12	0.13
Total Mass								<b>1.00</b>	<b>0.92</b>	0.06	-0.02	0.18	<b>0.63</b>	-0.05	<b>0.67</b>	-0.54	0.01	<b>0.68</b>	0.25	0.26
OTA Mass									<b>1.00</b>	-0.30	-0.07	0.34	0.42	-0.21	<b>0.77</b>	<b>-0.62</b>	0.12	0.55	0.32	-0.33
Spectral Range minimum										<b>1.00</b>	<b>0.93</b>	-0.59	-0.01	0.09	-0.57	0.07	0.27	0.21	0.26	0.29
Diff. Lim. λ											<b>1.00</b>	-0.48	-0.02	0.16	-0.49	0.46	0.44	0.33	0.39	0.14
Operating Temp.												<b>1.00</b>	0.42	0.03	<b>0.68</b>	-0.09	0.02	-0.28	0.06	-0.08
Avg. Input Power													<b>1.00</b>	0.51	0.56	0.05	0.38	0.15	0.42	0.19
Data Rate													<b>1.00</b>	0.01	<b>0.76</b>	<b>0.85</b>	0.20	<b>0.84</b>	0.26	
Design Life														<b>1.00</b>	-0.16	0.10	0.12	0.22	0.02	
TRL															<b>1.00</b>	0.89	-0.16	0.79	0.53	
Year of Dev.																<b>1.00</b>	0.11	<b>0.96</b>	0.21	
Dev. Period																	<b>1.00</b>	0.36	0.22	
Date of Launch																		<b>1.00</b>	0.24	
Orbit																			<b>1.00</b>	

Figure 1: Cross-Correlation Matrix of data base for 15 Space Telescope Systems. Correlations which are at least 95% significant are **Bolded**, e.g. for 12 data points a correlation of greater than 60% is significant to better than 95%.

A careful study of the cross-correlation matrix shows that OTA Cost is highly correlated with Aperture Diameter, Primary Mirror Focal Length, System Focal Length, Pointing Stability, Total Mass, OTA Mass, Design Life and Development Period. But, caution is required because not all of these variables are independent. Aperture Diameter correlates with PM Focal Length and System Focal Length, simply because larger aperture telescopes tend to have longer focal lengths. Also, because larger aperture telescopes have smaller point spread functions (or plate scales) they need to have smaller pointing stabilities, hence, the inverse correlation between aperture size and pointing. Obviously,

the larger the telescope aperture, the more massive the telescope will be. Additionally, the accompanying science instruments will undoubtedly be larger and more massive as well as the spacecraft. It is interesting to note that there does not appear to be a correlation between cost and operating temperature or diffraction limited performance. One explanation might be that they tend to cancel each other out. While the mirrors for longer wavelength systems are easier to manufacture, their cryogenic operating temperature increases cost. And, while visible systems operate at room temperature, their mirrors and structures are more difficult.

The second step is to select candidate CER variables and perform a regression analysis (Figure 2). The statistical indicators of ‘Goodness of Fit’ and ‘Significance’ are used to evaluate the results of this analysis. Goodness of Fit is tested via a range of statistical measures, including Pearson’s  $r^2$  coefficient, Student T-Test p-value and standard percent error (SPE). Pearson’s  $r^2$  (typically denoted as just  $r^2$ ) describes the percentage of agreement between the model and the actual cost. For multi-variable models, we use Adjusted Pearson’s  $r^2$  (or  $r^2_{adj}$ ) which accounts for the number of data points and the number of variables. In general, the closer  $r^2$  (or  $r^2_{adj}$ ) is to 1.0 or 100%, the better the model. SPE is a normalized standard deviation of the fit residual (difference between data and fit) to the fit. The closer SPE is to 0, the better the fit. Please note that since SPE is normalized, a small variation divided by a very small parameter coefficient can yield a very large SPE. The p-value is the probability that a fit or correlation would occur if the variables are independent of each other. The closer the p-value is to 0, the more significant the fit or correlation. The closer it is to 1, the less significant. If the p-value for a given variable is small, then removing it from the model would cause a large change to the model. If it is large, then removing the variable will have a negligible effect. Also, it is important to consider how many data points are included in a given correlation, fit or regression.

rev. 5.3.10		OTA Cost vs V1						
Variable Name		Aperture Diameter	PM F Len.	PM #	OTA Volume	FOV	Pointing Stability	OTA Mass
Var.	p-value	1.75 0.00	1.74 0.00	1.08 0.34	0.65 0.00	-0.15 0.68	-0.76 0.01	0.97 0.00
Adjusted $r^2$	67%	75%	2%	72%	4%	33%	56%	
SPE	146%	117%	729%	116%	719%	205%	88%	
n	15	13	13	13	14	8	12	

Variable Name		OTA Area Density	Spectral Range minimum	Diff. Lim. $\lambda$	Operating Temp.	Year of Dev. (exp)	Date of Launch (exp)
Var.	p-value	0.16 0.83	-0.16 0.46	-0.19 0.42	0.27 0.43	0.00 0.98	0.04 0.42
Adjusted $r^2$	-9%	2%	-1%	-7%	-8%	-3%	
SPE	1084%	900%	845%	1162%	1225%	1183%	
n	12	15	13	14	14	15	

Figure 2: Single Variable Regression Analysis for OTA Cost

#### 4. SINGLE-VARIABLE MODELS

From an engineering and a scientific perspective, aperture is the best parameter with which to build a space telescope cost model. Aperture defines the observatory’s science performance (sensitivity and resolution) and determines the payload’s size and mass. As discussed in Section 3, Aperture Diameter correlates with all of the other variables which significantly correlate with cost. From Figure 2, OTA cost varies as a function of diameter according to:

$$\text{OTA Cost} \sim \text{Diameter}^{1.75} \quad (N = 15; r^2 = 67\%; \text{SPE} = 146)$$

This CER is based on 15 data points, one of which (SOFIA) is not actually a space telescope. But as shown in Figure 3, SOFIA’s OTA cost is ‘in-family’ for its aperture. And, removing SOFIA from the regression has a negligible effect:

$$\text{OTA Cost} \sim \text{Diameter}^{1.8} \quad (N = 14; r^2 = 67\%; \text{SPE} = 142)$$

However, both CERs only account for 67% of the cost variation and both are noisy. Therefore, a single variable aperture diameter model is not a good CER. Other variables are needed to account for the remaining 33% of cost variation.

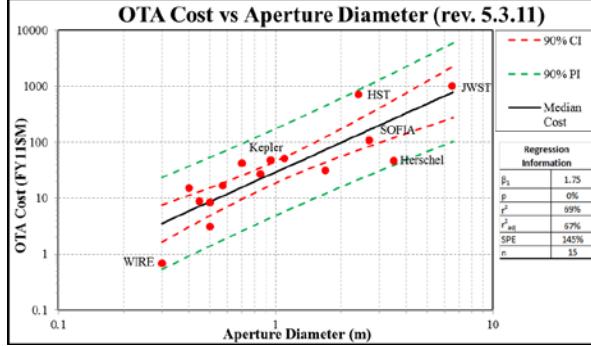


Figure 3: OTA Cost vs Aperture Diameter

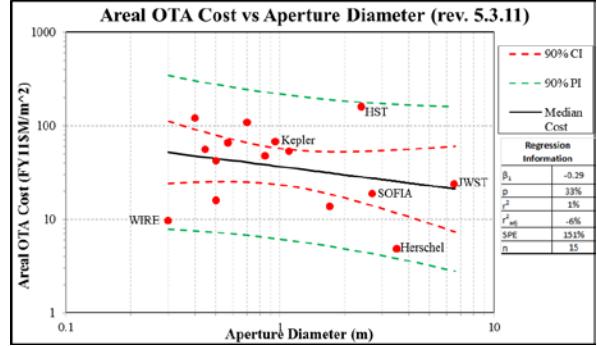


Figure 4: OTA Areal Cost vs Aperture Diameter

One concern about cost versus diameter is that JWST drives the fit. As a simple sanity check, the data was normalized by collecting area to define Areal Cost (Figure 4). By eliminating the diameter influence, data spread associated with second order factors such as wavelength or operational temperature can be identified. The key point of Figure 4 is that areal cost decreases as a function of aperture diameter. Thus, given that the number of collected photons is proportional to collecting area, larger aperture telescopes have a greater return on investment (ROI) than smaller aperture telescopes.

While an Aperture based CER may be most logical for an optical engineer, many believe that Mass is the more important CER. Total system mass determines what vehicle can be used to launch the mission. And, significant engineering costs are expended to keep a given payload inside of its allocated mass budget, for example: light-weighting mirrors and structure. It is factual to assert that space telescopes are designed to meet a specific mass budget. From Figure 2, OTA cost varies as a function of mass according to:

$$\text{OTA Cost} \sim \text{OTA Mass}^{0.97} \quad (N = 12; r^2 = 56\%; \text{SPE} = 88)$$

Compared to the Aperture Diameter CER, OTA Mass is less noisy. But, it still only accounts for 56% of the cost variation. One problem is that this CER is based on 12 data points, one of which (SOFIA) is not actually a space telescope. SOFIA is actually an ‘attached’ telescope. It flies attached to a 747 aircraft. And, by flying on a 747 aircraft, it can be designed to an entirely different mass budget paradigm. But, as shown in Figures 3 and 5, while SOFIA has the approximately the same aperture size and OTA mass as HST; it has a significantly different cost than HST. One explanation might be because SOFIA has a longer diffraction limited wavelength but still operates close to ambient temperature. Another explanation might be that SOFIA does not have the challenges of operating in space, i.e. in vacuum or micro-gravity. Removing SOFIA from the regression has a significant effect:

$$\text{OTA Cost} \sim \text{OTA Mass}^{1.1} \quad (N = 11; r^2 = 96\%; \text{SPE} = 78\%)$$

Without SOFIA, the OTA Mass CER accounts for 96% of the cost variation and has a smaller standard percent error.

Another potential wavelength story is Herschel and Kepler. Herschel and Kepler have essentially the same mass and cost, but vastly different apertures, diffraction limits and operating temperatures. Based only on aperture, Herschel should be more expensive than Kepler, but it has a significantly longer diffraction limit and lower operating temperature.

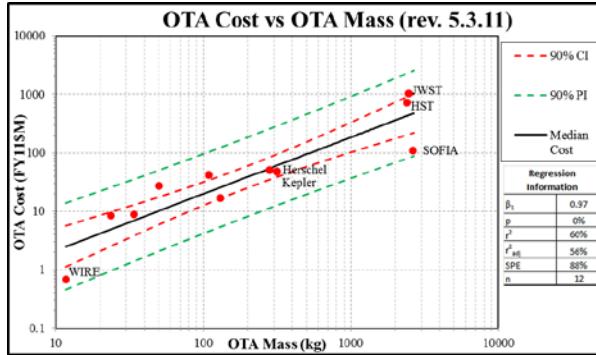


Figure 5: OTA Cost vs OTA Mass

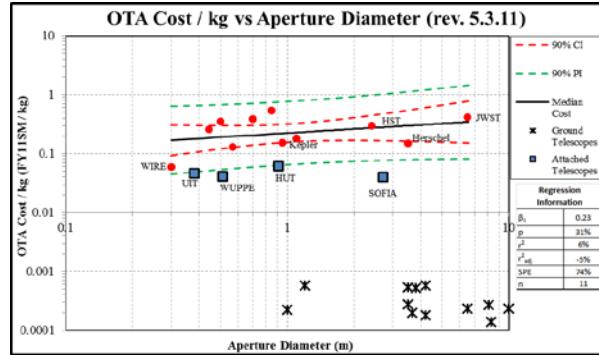


Figure 6: OTA Cost Density vs Aperture Diameter

A very interesting tool for analyzing the role of mass on cost is cost density (cost per kg). Figure 6 plots OTA cost per kg versus OTA aperture diameter. Several obvious conclusions can be drawn from Figure 6. First, all free flying space telescopes have approximately the same cost per kg – independent of aperture diameter. And, all ground telescopes also have approximately the same cost per kg – independent of aperture diameter. And, space telescopes cost about 1000X per kg more than ground telescopes – independent of aperture diameter. Second, while the conclusion regarding attached telescopes might appear to be equally obvious (that attached telescopes are approximately 5X less expensive than free-flying), it is not. For the three shuttle missions (UIT, WUPPE and HUT), the data base only has ‘instrument’ data, i.e. cost and mass of the telescope and science instruments (detectors, electronics, etc.). More research is required. Thus, the only conclusion which can be drawn from Figure 6 is that it costs more per kg to make low areal density flight telescopes than it costs to make massive ground telescopes. One explanation for this data might be that each of these mission ‘types’ are built to different design rules. While all three types need similar wavefront shape and pointing stabilities as a function of aperture diameter, they have different static gravity and dynamic jitter environments; and different mass budgets for achieving the required wavefront shape and pointing stability.

A final caution about using mass as a CER can be found by considering HST vs JWST. While the HST and JWST OTAs have similar mass and cost (although JWST is a bit more expensive), this relationship does not hold at the system level. HST system mass is nearly 2X more than JWST, yet JWST is slightly more expensive than HST.

## 5. CONCLUSIONS

Cost models are invaluable for system designers. They identify major architectural cost drivers and allow high-level design trades. They enable cost-benefit analysis for technology development investment. And, they provide a basis for estimating total project cost. Cost and engineering data has been collected on 59 different parameters for 39 x-ray, UV, optical, infrared, microwave and radio space telescopes. But to date, only the 29 normal-incidence UV, Optical, Infrared (UVOIR) missions have been studied for cost modeling. And, of these 29, sufficient data exists for only 15 with which to develop an Optical Telescope Assembly (OTA) cost model. Statistical correlations have been evaluated between 18 of 59 variables. And, these parameters have been used to develop single variable cost estimating relationships (CERs) which are evaluated for their ‘goodness’.

From an engineering and a scientific perspective, aperture is the best parameter to build a space telescope cost model. Aperture defines the observatory’s science performance (sensitivity and resolution) and determines the payload’s size and mass. OTA cost varies as a function of diameter according to:

$$\text{OTA Cost} \sim \text{Diameter}^{1.75} \quad (N = 15; r^2 = 67\%; SPE = 146)$$

This CER is based on 15 data points, one of which (SOFIA) is not actually a space telescope. If SOFIA is removed from the regression it has a negligible effect:

$$\text{OTA Cost} \sim \text{Diameter}^{1.8} \quad (N = 14; r^2 = 67\%; SPE = 142)$$

However, both CERs only account for 67% of the cost variation and both are noisy. Therefore, a single variable aperture diameter model is not a good CER. Other variables are needed to account for the remaining 33% of cost variation.

A key point of the aperture model is that the diameter coefficient is less than 2. Therefore, areal cost decreases as a function of aperture diameter. Hence, given that the number of photons collected is proportional to collecting area, larger aperture telescopes have a greater return on investment (ROI) than smaller aperture telescopes.

While an Aperture based CER may be most logical for an optical engineer, Mass may be a more important CER. Total system mass determines what vehicle can be used to launch a mission. And, significant engineering costs are expended to keep a given payload inside of its allocated mass budget. OTA cost varies as a function of mass according to:

$$\text{OTA Cost} \sim \text{OTA Mass}^{0.97} \quad (N = 12; r^2 = 56\%; SPE = 88)$$

Compared to the Aperture Diameter CER, OTA Mass is less noisy. But, it still only accounts for 56% of the cost variation. However, this CER is based on 12 data points, one of which (SOFIA) is not actually a space telescope. Removing SOFIA from the regression has a significant effect:

$$\text{OTA Cost} \sim \text{OTA Mass}^{1.1} \quad (N = 11; r^2 = 96\%; SPE = 78\%)$$

Without SOFIA, the OTA Mass CER accounts for 96% of the cost variation and has a smaller standard percent error.

Finally, cost density (cost per kg) as a function of aperture diameter indicates that for given ‘classes’ of telescopes of missions, the cost per kg is independent of aperture diameter. Thus, it costs more per kg to make low areal density flight telescopes than it costs to make massive ground telescopes. One explanation might be that it requires significantly more ‘engineering’ effort to design a low areal density telescope with the required wavefront shape and pointing stability for its operational (static gravity load and dynamic jitter) environment than it does for a more massive telescope.

## REFERENCES

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# Agenda

- Introduction and Summary
- Data Collection Methodology
- Statistical Analysis Methodology
- What to Model?: OTA or Total Mission Cost
- Single Variable Modes: Mass and Diameter
- Multi-Variable Models
- Conclusions



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## Parametric Cost Models

Parametric cost models have several uses:

- high level mission concept design studies,
- identify major architectural cost drivers,
- allow high-level design trades,
- enable cost-benefit analysis for technology development investment, and
- provide a basis for estimating total project cost.



However

All Cost Models are Wrong!

But Some are Useful.

The Rest will get you into Trouble.



## DISCLAIMER

Cost Models are only as good as their Data Base

This is a work in progress.

The results evolve as we add new missions to the Database, add data to or correct data in the Database.



## Findings

Aperture Diameter is principle cost driver for space telescopes.

$$\text{OTA Cost} \sim \text{Diameter}^{1.8}$$

$$\text{OTA Cost} \sim \text{Dia}^{1.6} \lambda^{-0.25}$$

Larger diameter OTAs cost less per square meter of aperture.

Longer wavelength OTAs cost less.

If all parameters are held constant, adding mass reduces cost & reducing mass increases cost.

Still examining Year of Development



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# Methodology

Data accumulated on 59 engineering and programmatic variables

18 Variables studied for Cost Estimating Relationships (CERs)

Data sources :

NAFCOM (NASA/ Air Force Cost Model) database,  
NICM (NASA Instrument Cost Model),  
NSCKN (NASA Safety Center Knowledge Now),  
RSIC (Redstone Scientific Information Center),  
REDSTAR (Resource Data Storage and Retrieval System),  
SICM (Scientific Instrument Cost Model),  
project websites, and interviews.



# Cost & Mass Definitions

## **Total Mission:**

- Spacecraft
- Science Instruments
- Telescope

## **Instrument:**

- Entire payload or experiment including telescope

## **Optical Telescope Assembly (OTA):**

- Primary mirror
- Secondary (and tertiary if appropriate) mirror(s)
- Support structure
- Mechanisms (actuators, etc.), Electronics, Software, etc.
- Assembly, Integration & Test



# Cost & Mass Definitions (2)

## Cost includes:

- Phase A-D (design, development, integration and test)

## Cost excludes:

- Pre-phase A (formulation)
- Phase E (launch/post-launch)
- Government labor costs (NASA employees: CS or support contractors)
- Government Furnished Equipment (GFE)
- Existing Contractor infrastructure which is not 'billed' to contract.
- These are 'First Unit' Costs only – no HST Servicing & there are no 2<sup>nd</sup> Systems.

## Mass includes:

- Dry mass only (no propellant)



# Technical Variables

Aperture Diameter

PM Focal Length

System Focal Length

Field of View

Pointing Stability

OTA Mass

Total Mass

Spectral Range Minimum

Wavelength of Diffraction Limit

Operating Temperature

Average Input Power

Data Rate

Design Life

Orbit



# Programmatic Variables

TRL (Technology Readiness Level)

Year of Development (or Start of Development)

Development Period

Launch Year



# Missions (8.6.11 Database)

Currently 45 missions in data base

33 ‘normal-incidence’ UVOIR and Infrared telescopes

5 grazing incidence X-Ray

7 Radio/Microwave

Data for microwave, radio wave & grazing incidence X-Ray/EUV provides wavelength diversity

To date only normal-incidence UVOIR and Microwave telescopes used for cost modeling

Cost Model Missions Database (8.6.11)	
<u>X-Ray Telescope</u>	<u>Infrared Telescopes</u>
Chandra (AXAF)	CALIPSO
Einstein (HERO-2)	Herschel
EUVE	IRAS
FOXSI	ISO
HERO	JWST
	SOFIA
	Spitzer (SIRTF)
	TRACE
	WIRE
	WISE
<u>UV/Optical Telescopes</u>	<u>Microwave Telescopes</u>
Commercial #1	ACTS
Commercial #2	Cloud SAT
Copernicus (OAO-3/PEP)	Planck
EO-1/ALI	WMAP
EUVE	
FUSE	
GALEX	
HST	
HUT	
ICESat	
IUE	
Kepler	
LANDSAT-7	
LRO/LROC NAC	
MO/MOC	
MO/MOLA	
MRO/HiRISE	
OAO-B/GEP	
SDO/AIA	
SOHO/EIT	
STEREO/SECCHI	
UIT	
WUPPE	
	<u>Radio Antenna</u>
	SWAS
	TDRS-1
	TDRS-7



# Missions (8.6.11 Database)

Of 37 ‘normal-incidence’ UVOIR and Microwave telescopes

27 are ‘Free Flying’

4 are ‘Attached’ and

5 are ‘Planetary/Other’

Additionally, some of these are Imaging and others are Spectroscopic.

We have not yet investigated the impact of this distinction, but expect spectroscopic to be lower cost.

Normal Incidence Database (8.6.11)	
<u>Free Flying Telescopes</u>	<u>Attached Telescopes</u>
ACTS	HUT
CALIPSO	SOFIA
Cloud SAT	UIT
Commercial #1	WUPPE
Commercial #2	
Copernicus (OAO-3/PEP)	
EO-1/ALI	
EUVE	LRO/LROC NAC
FUSE	MO/MOC
GALEX	MO/MOLA
Herschel	MRO/HiRISE
HST	STEREO/SECCHI
ICESat	
IRAS	
ISO	
IUE	
JWST	
Kepler	
LANDSAT-7	
OAO-B/GEP	
Planck	
SDO/AIA	
SOHO/EIT	
Spitzer (SIRTF)	
TRACE	
WIRE	
WISE	
WMAP	



# Missions (8.6.11 Database)

For some have only Mission data  
and for others have both OTA and  
Mission data.

We have OTA Cost:

& Diameter data for 15  
& Mass data for 13

Cost Model Variables for Free Flying UVO/IR Systems (rev. 11.6.10)	
Parameter	% of data
Total Cost	89%
OTA Cost	46%
Total Cost & OTA Cost	68%
Aperture Diameter	100%
PM F Len.	71%
System F Len.	89%
FOV	86%
Pointing Stability	39%
Total Mass	93%
OTA Mass	86%
Spectral Range minimum	96%
Diffraction Limited Wavelength	61%
Operating Temperature	93%
Avg. Input Power	89%
Data Rate	79%
Design Life	96%
TRL	32%
Year of Dev.	93%
Dev. Period	89%
Date of Launch	96%
Orbit	82%
Average	78%



## Missions (8.6.11 Database)

These are the missions used in our cost model analysis

15 are ‘Free Flying’

4 is ‘Attached’ and

1 is ‘Planetary’

Of these 8 are spectroscopic or non-imaging.

Normal Incidence Database (8.6.11)	
<u>Free Flying Telescopes</u>	<u>Attached Telescopes</u>
Cloud SAT	HUT
Commercial #1	SOFIA
Commercial #2	UIT
Copernicus (OAO-3/PEP)	WUPPE
GALEX	
Herschel	
HST	
IRAS	
JWST	
Kepler	
OAO-B/GEP	
Planck	
Spitzer (SIRTF)	
WIRE	
WISE	
<u>Planetary Telescopes</u>	
MRO/HiRISE	



# Agenda

- Introduction and Summary
- Data Collection Methodology
- **Statistical Analysis Methodology**
  - What to Model?: OTA or Total Mission Cost
  - Single Variable Modes: Mass and Diameter
  - Multi-Variable Models
  - Conclusions



# Model Creation

Start with Correlation Matrix.

Look for Variables which are Highly Correlated with Cost.

The higher the correlation the greater the Cost Variation which is explained by a given Variable.

Sign of correlation is important and must be consistent with Engineering Judgment.

Important for Multi-Variable Models:

We want Variables which Independently effect Cost.

When Variables ‘cross-talk’ with each other it is called Multi-Collinearity.

Thus, avoid Variables which are highly correlated with each other.



# Goodness of Correlation, Fits and Regressions

‘Correlation’ between variables and ‘Goodness’ of single variable models is evaluated via Pearson’s  $r^2$  standard percent error (SPE), and Student’s T-Test p-value.

‘Goodness’ of multivariable fits are evaluated via Pearson’s Adjusted  $r^2$  which accounts for number of data points and number of variables.

Pearson’s  $r^2$  coefficient describes the percentage of agreement between the fitted values and the actual data.

The closer  $r^2$  is to 1, the better the fit.

SPE is a normalized standard deviation of the fit residual (difference between data and fit) to the fit.

The closer SPE is to 0, the better the fit



## Significance

The final issue is whether or not a correlation or fit is significant.

p-value is the probability that the fit or correlation would occur if the variables are independent of each other.

The closer p-value is to 0, the more significant the fit or correlation.

The closer p-value is to 1, the less significant.

If the p-value for a given variable is small, then removing it from the model would cause a large change to the model.

If p-value is large, then removing the variable will have a negligible effect

It is only possible to ‘test’ if the correlation between two variables is significant.

It is not possible to ‘test’ if two variables are independent.



## Cross-Correlation Matrix



# Cross-Correlation Matrix

rev. 8.1.11	Total Cost	OTA Cost	Aperture Diameter	PM F Len.	System F Len.	FOV	Pointing Stability	Total Mass	OTA Mass	Spectral Range minimum	Diff. Lim. λ	Operating Temp.	Total Avg. Input Power	Data Rate	Design Life	TRL	Year of Dev.	Dev. Period	Date of Launch	Orbit
units	(FY11\$M)	(FY11\$M)	(m)	(m)	(m)	(°)	(Arc-Sec / Sec)	(kg)	(kg)	(nm)	(E)	(E)	(Watts)	(Kbps)	(months)	TRL	(year)	(months)	(year)	(km)
Total Cost	<b>1.00</b>	<b>0.85</b>	<b>0.69</b>	0.21	0.52	0.13	-0.72	0.68	<b>0.85</b>	0.21	-0.05	-0.11	0.57	0.05	0.30	-0.45	-0.15	0.70	0.03	0.46
OTA Cost		<b>1.00</b>	<b>0.78</b>	<b>0.88</b>	<b>0.72</b>	-0.13	-0.80	0.84	<b>0.95</b>	-0.16	-0.20	0.04	0.37	0.22	<b>0.64</b>	-0.61	-0.03	<b>0.62</b>	0.16	0.07
Aperture Diameter			<b>1.00</b>	0.36	<b>0.72</b>	-0.04	-0.71	0.59	<b>0.84</b>	0.51	0.46	-0.05	0.44	-0.10	0.43	-0.28	-0.08	0.45	0.05	0.05
PM F Len.				<b>1.00</b>	0.70	0.13	-0.77	0.69	<b>0.89</b>	-0.48	-0.38	0.18	0.21	0.04	0.56	-0.35	-0.11	0.41	0.11	0.00
System F Len.					<b>1.00</b>	-0.29	-0.47	<b>0.61</b>	<b>0.70</b>	0.02	-0.13	-0.04	0.13	-0.16	0.60	-0.38	-0.22	0.43	-0.03	0.18
FOV						<b>1.00</b>	-0.33	0.03	0.27	0.24	0.26	-0.12	0.10	0.19	-0.15	-0.31	0.19	-0.01	0.18	0.06
Pointing Stability							<b>1.00</b>	<b>-0.62</b>	<b>-0.87</b>	0.16	0.18	-0.13	-0.46	-0.03	-0.54	0.26	-0.04	<b>-0.63</b>	-0.24	-0.03
Total Mass								<b>1.00</b>	<b>0.92</b>	-0.10	0.10	0.01	0.39	-0.19	0.46	-0.54	-0.31	0.51	-0.16	0.24
OTA Mass									<b>1.00</b>	-0.22	-0.22	0.00	0.27	-0.17	0.41	<b>-0.62</b>	-0.04	<b>0.69</b>	0.18	-0.35
Spectral Range minimum										<b>1.00</b>	<b>0.96</b>	-0.24	0.16	0.10	0.06	-0.09	0.19	0.17	0.19	0.12
Diff. Lim. λ											<b>1.00</b>	-0.28	0.10	0.15	-0.20	0.12	0.35	0.23	0.32	0.26
Operating Temp.												<b>1.00</b>	0.12	-0.05	0.27	0.11	-0.06	-0.39	-0.09	-0.06
Total Avg. Input Power													<b>1.00</b>	0.50	0.57	0.13	0.59	0.06	0.57	0.25
Data Rate														<b>1.00</b>	0.14	0.63	0.72	-0.09	0.70	0.28
Design Life															<b>1.00</b>	-0.15	0.12	0.15	0.25	0.33
TRL																<b>1.00</b>	0.67	-0.17	0.64	0.32
Year of Dev.																	<b>1.00</b>	-0.13	0.97	0.22
Dev. Period																		<b>1.00</b>	0.09	0.36
Date of Launch																			<b>1.00</b>	0.28
Orbit																				<b>1.00</b>

Correlations which are at least 95% significant are **Bolded**, e.g. for 12 data points a correlation of greater than 60% is significant to better than 95%.



# Cross-Correlation Matrix

rev. 8.1.11	Total Cost	OTA Cost	Aperture Diameter	PM F Len.	System F Len.	FOV	Pointing Stability (Arc-Sec / Sec)	Total Mass	OTA Mass	Spectral Range minimum	Diff. Lim. λ	Operating Temp.	Total Avg. Input Power	Data Rate	Design Life	TRL	Year of Dev.	Dev. Period	Date of Launch	Orbit
units	(FY11\$M)	(FY11\$M)	(m)	(m)	(m)	(°)		(kg)	(kg)	(Å)	(Å)	(K)	(Watts)	(Kbps)	(months)	TRL	(year)	(months)	(year)	(km)
Total Cost	1.00	0.85	0.69	0.21	0.52	0.13	-0.72	0.68	0.85	0.21	-0.05	-0.11	0.57	0.05	0.30	-0.45	-0.15	0.70	0.03	0.46
OTA Cost		1.00	0.78	0.88	0.72	-0.13	-0.80	0.84	0.95	-0.16	-0.20	0.04	0.37	0.22	0.64	-0.61	-0.03	0.62	0.16	0.07
Aperture Diameter			1.00	0.36	0.72	-0.04	-0.71	0.59	0.84	0.51	0.46	-0.05	0.44	-0.10	0.43	-0.28	-0.08	0.45	0.05	0.05
PM F Len.																		11	0.00	
System F Len.																		03	0.18	
FOV																		18	0.06	
Pointing Stability																		24	-0.03	
Total Mass																		16	0.24	
OTA Mass																		18	-0.35	
Spectral Range minimum																		19	0.12	
Diff. Lim. λ																		32	0.26	
Operating Temp.																		09	-0.06	
Total Avg. Input Power																		57	0.25	
Data Rate																		70	0.28	
Design Life																		25	0.33	
TRL																		54	0.32	
Year of Dev.																		97	0.22	
Dev. Period																		09	0.36	
Date of Launch																		00	0.28	
Orbit																			1.00	

OTA Cost has significant correlations with:

- Aperture Diameter
- Primary Mirror & System Focal Length (Volume)
- Pointing Stability (inverse correlation)
- OTA Mass
- Design Life

Total Cost has significant correlations with:

- Aperture Diameter
- Primary Mirror & System Focal Length (Volume)
- Pointing Stability (inverse correlation)
- OTA & Total Mass
- Average Power
- Design Life
- Development Period

No correlation for wavelength or temperature  
TRL correlation is ‘weak’



# Not all Correlated Variables are Independent

rev. 8.1.11	Total Cost	OTA Cost	Aperture Diameter	PM F Len.	System F Len.	FOV	Pointing Stability	Total Mass	OTA Mass	Spectral Range minimum	Diff. Lim. λ	Operating Temp.	Total Avg. Input Power	Data Rate	Design Life	TRL	Year of Dev.	Dev. Period	Date of Launch	Orbit
units	(FY11\$M)	(FY11\$M)	(m)	(m)	(m)	(°)	(Arc-Sec / Sec)	(kg)	(kg)	(Å)	(Å)	(K)	(Watts)	(Kbps)	(months)	TRL	(year)	(months)	(year)	(km)
Total Cost	1.00	0.85	0.69	0.21	0.52	0.13	-0.72	0.68	0.85	0.21	-0.05	-0.11	0.57	0.05	0.30	-0.45	-0.15	0.70	0.03	0.46
OTA Cost		1.00	0.78	0.88	0.72	-0.13	-0.80	0.84	0.95	-0.16	-0.20	0.04	0.37	0.22	0.64	-0.61	-0.03	0.62	0.16	0.07
Aperture Diameter			1.00	0.36	0.72	-0.04	-0.71	0.59	0.84	0.51	0.46	-0.05	0.44	-0.10	0.43	-0.28	-0.06	0.45	0.05	0.05
PM F Len.				1.00	0.70	0.15	-0.77	0.69	0.89	-0.48	-0.38	0.18	0.21	0.04	0.56	-0.35	-0.11	0.41	0.11	0.00
System F Len.					1.00	-0.29	-0.47	0.61	0.70	0.02	-0.13	-0.04	0.13	-0.16	0.60	-0.38	-0.22	0.43	-0.03	0.18
FOV						1.00	-0.33	0.03	0.27	0.24	0.26	-0.12	0.10	0.19	-0.15	-0.31	0.19	-0.01	0.18	0.06
Pointing Stability																		4	-0.03	
Total Mass																		6	0.24	
OTA Mass																		8	-0.35	
Spectral Range minimum																		9	0.12	
Diff. Lim. λ																		2	0.26	
Operating Temp.																		9	-0.06	
Total Avg. Input Power																		7	0.25	
Data Rate																		9	0.28	
Design Life																		5	0.33	
TRL																		4	0.32	
Year of Dev.																		7	0.22	
Dev. Period																		9	0.36	
Date of Launch																		9	0.28	
Orbit																			1.00	

Larger Diameter OTAs:

- have longer Focal Lengths
- have smaller Pointing Stability Requirements
- are just bigger and thus more Massive
- have larger instruments with are more Massive & require Power
- require bigger spacecraft which are more Massive & require Power
- need a long Design Life
- take longer to Develop
- are more Recent – older OTAs were smaller

All these variable are dependent on Aperture Diameter (co-linear).



# Variable Linkages

rev. 8.1.11	Total Cost	OTA Cost	Aperture Diameter	PM F Len.	System F Len.	FOV	Pointing Stability	Total Mass	OTA Mass	Spectral Range minimum	Diff. Lim. $\lambda$	Operating Temp.	Total Avg. Input Power	Data Rate	Design Life	TRL	Year of Dev.	Dev. Period	Date of Launch	Orbit
units	(FY11SM)	(FY11SM)	(m)	(m)	(m)	(°)	(Arc-Sec / Sec)	(kg)	(kg)	(nm)	(e)	(K)	(Watts)	(Kbps)	(months)	TRL	(year)	(months)	(year)	(km)
Total Cost	1.00	0.85	0.69	0.21	0.52	0.13	-0.72	0.68	0.85	0.21	-0.05	-0.11	0.57	0.05	0.30	-0.45	-0.15	0.70	0.03	0.46
OTA Cost		1.00	0.78	0.88	0.72	-0.13	-0.80	0.84	0.95	-0.16	-0.20	0.04	0.37	0.22	0.64	-0.61	-0.03	0.62	0.16	0.07
Aperture Diameter			1.00	0.36	0.72	-0.04	-0.71	0.59	0.84	0.51	0.46	-0.05	0.44	-0.10	0.43	-0.28	-0.08	0.45	0.05	0.05
PM F Len.				1.00	0.70	0.15	-0.77	0.69	0.89	-0.48	-0.38	0.18	0.21	0.04	0.56	-0.55	-0.11	0.41	0.11	0.00
System F Len.					1.00	-0.29	-0.47	0.61	0.70	0.02	-0.13	-0.04	0.13	-0.16	0.60	-0.38	-0.22	0.43	-0.03	0.18
FOV						1.00	-0.33	0.03	0.27	0.24	0.26	-0.12	0.10	0.19	-0.15	-0.31	0.19	-0.01	0.18	0.06
Pointing Stability							1.00	-0.62	-0.87	0.16	0.18	-0.13	-0.46	-0.03	-0.54	0.26	-0.04	-0.63	-0.24	-0.03
Total Mass								1.00	0.92	-0.10	0.10	0.01	0.39	-0.19	0.46	-0.54	-0.31	0.51	-0.16	0.24
OTA Mass									1.00	-0.22	-0.22	0.00	0.27	-0.17	0.41	-0.62	-0.04	0.69	0.18	-0.35
Spectral Range minimum																			0.12	
Diff. Lim. $\lambda$																			0.26	
Operating Temp.																			-0.06	
Total Avg. Input Power																			0.25	
Data Rate																			0.28	
Design Life																			0.33	
TRL																			0.32	
Year of Dev.																			0.22	
Dev. Period																			0.36	
Date of Launch																			0.28	
Orbit																			1.00	

Correlation Matrix can be used to identify variable cross-linkages which should be reconciled with Engineering Judgment.

Aperture Diameter and Pointing Stability have a large negative correlation: Larger Diameter OTAs required smaller Pointing Stability.

Pointing Stability and OTA Mass have a large negative correlation: Small Pointing Stability requires a very stiff, i.e. Massive, OTA.



# Wavelength and Temperature

rev. 8.1.11	Total Cost	OTA Cost	Aperture Diameter	PM F Len.	System F Len.	FOV	Pointing Stability	Total Mass	OTA Mass	Spectral Range minimum	Diff. Lim. λ	Operating Temp.	Total Avg. Input Power	Data Rate	Design Life	TRL	Year of Dev.	Dev. Period	Date of Launch	Orbit
units	(FY11SM)	(FY11SM)	(m)	(m)	(m)	(°)	(Arc-Sec / Sec)	(kg)	(kg)	(nm)	(E)	(E)	(Watts)	(Kbps)	(months)	TRL	(year)	(months)	(year)	(km)
Total Cost	1.00	0.85	0.69	0.21	0.52	0.13	-0.72	0.68	0.85	0.21	-0.05	-0.11	0.57	0.05	0.30	-0.45	-0.15	0.70	0.03	0.46
OTA Cost		1.00	0.78	0.88	0.72	-0.13	-0.80	0.84	0.95	-0.16	-0.20	0.04	0.37	0.22	0.64	-0.61	-0.03	0.62	0.16	0.07
Aperture Diameter			1.00	0.36	0.72	-0.04	-0.71	0.59	0.84	0.51	0.46	-0.05	0.44	-0.10	0.43	-0.28	-0.08	0.45	0.05	0.05
PM F Len.				1.00	0.70	0.13	-0.77	0.69	0.89	-0.48	-0.38	0.18	0.21	0.04	0.56	-0.35	-0.11	0.41	0.11	0.00
System F Len.					1.00	-0.29	-0.47	0.61	0.70	0.02	-0.13	-0.04	0.13	-0.16	0.60	-0.38	-0.22	0.43	-0.03	0.18
FOV						1.00	-0.33	0.03	0.27	0.24	0.26	-0.12	0.10	0.19	-0.15	-0.31	0.19	-0.01	0.18	0.06
Pointing Stability							1.00	-0.62	-0.87	0.16	0.18	-0.13	-0.46	-0.03	-0.54	0.26	-0.04	-0.63	-0.24	-0.03
Total Mass								1.00	0.92	-0.10	0.10	0.01	0.39	-0.19	0.46	-0.54	-0.31	0.51	-0.16	0.24
OTA Mass									1.00	-0.22	-0.22	0.00	0.27	-0.17	0.41	-0.62	-0.04	0.69	0.18	-0.35
Spectral Range minimum										1.00	0.96	-0.24	0.16	0.10	0.06	-0.09	0.19	0.17	0.19	0.12
Diff. Lim. λ											1.00	-0.28	0.10	0.15	-0.20	0.12	0.35	0.23	0.32	0.26
Operating Temp.												1.00	0.12	-0.05	0.27	0.11	-0.06	-0.39	-0.09	-0.06
Total Avg. Input Power																				
Data Rate																				
Design Life																				
TRL																				
Year of Dev.																				
Dev. Period																				
Date of Launch																				
Orbit																				

As expected Spectral Range and  
Diffraction Limit are highly correlated.  
Operating Temperature are inversely correlated.

But neither are significantly correlated with Cost – probably  
because they cancel either other out.



# Agenda

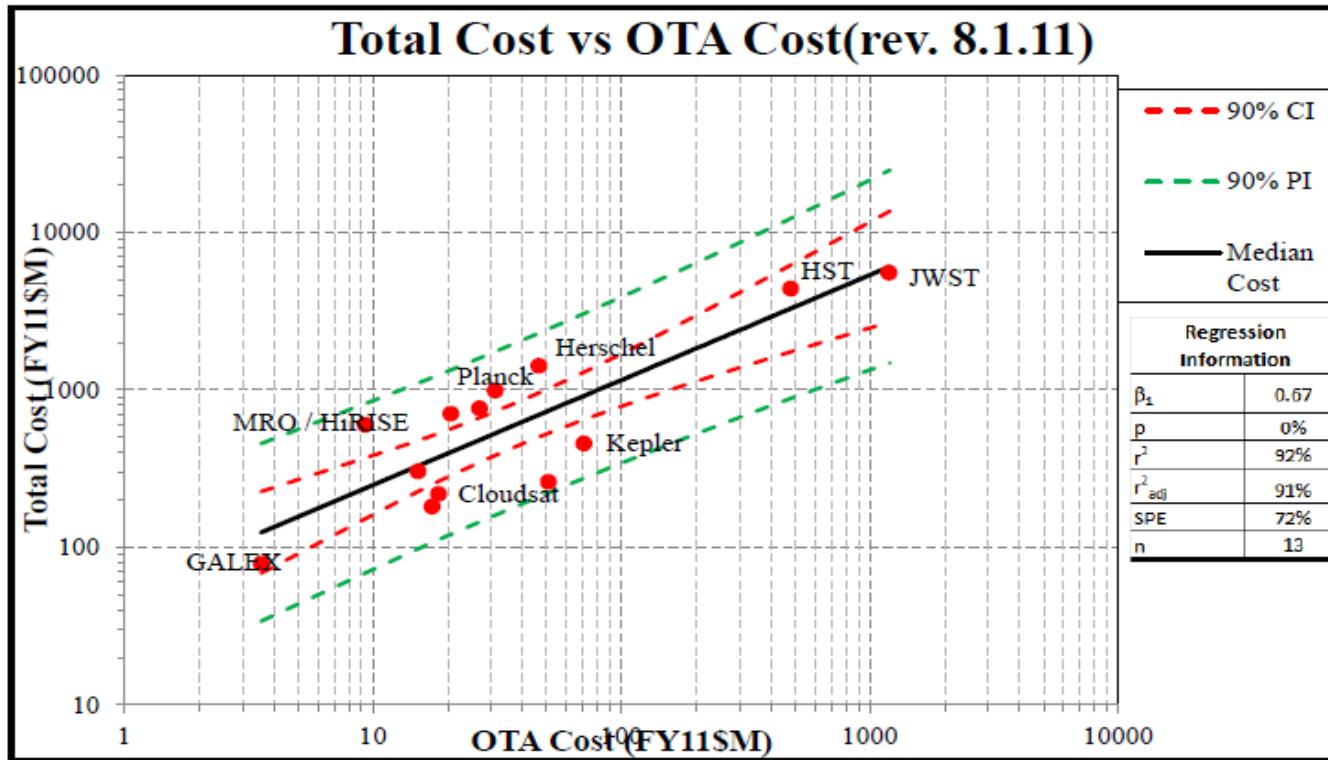
- Introduction and Summary
- Data Collection Methodology
- Statistical Analysis Methodology
- **What to Model?: OTA or Total Mission Cost**
- Single Variable Modes: Mass and Diameter
- Multi-Variable Models
- Conclusions



# OTA Cost or Total Cost

Engineering judgment says that OTA cost is most closely related to OTA engineering parameters.

But, managers and mission planners are really more interested in total Phase A-D cost.

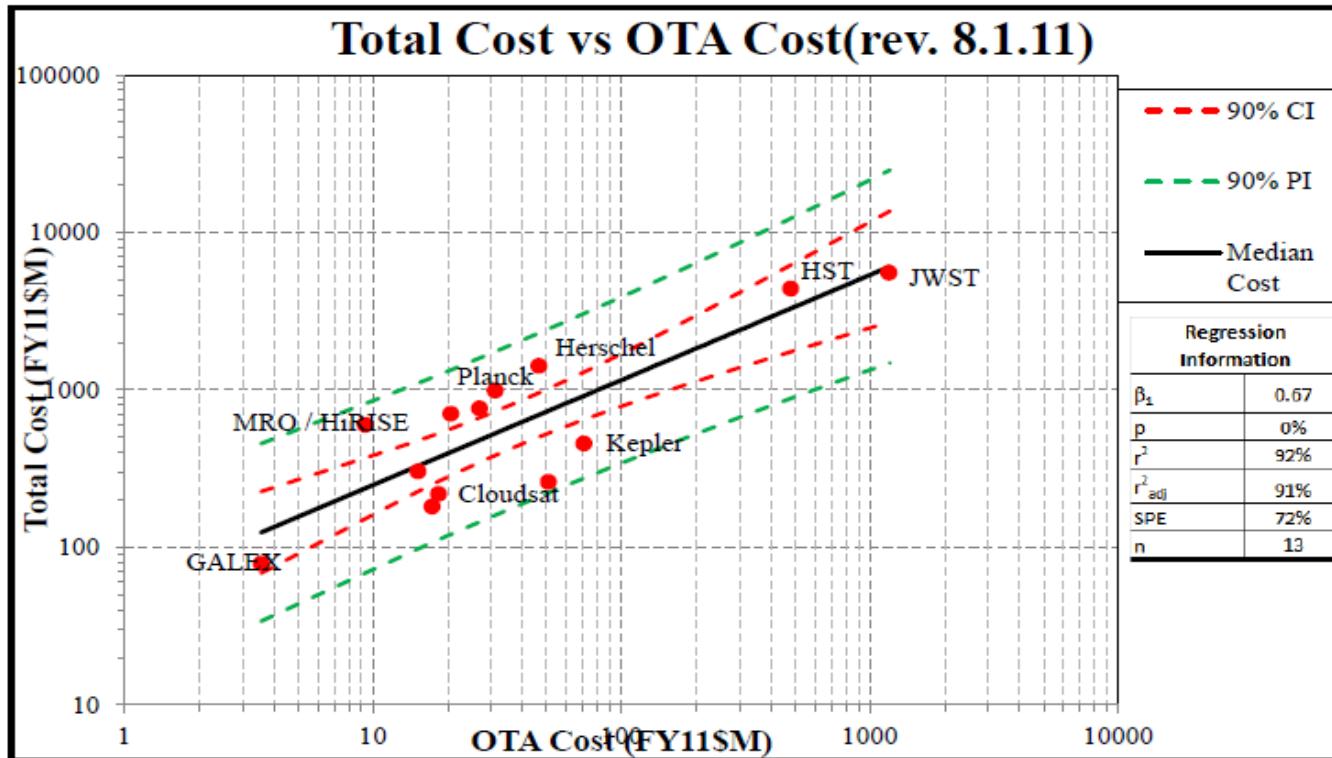




# OTA Cost or Total Cost

Given that Total Cost tracks closely with OTA Cost, and that I'm an optics person and have accumulated mostly OTA data.

Our primary emphasis is to develop an OTA cost model.



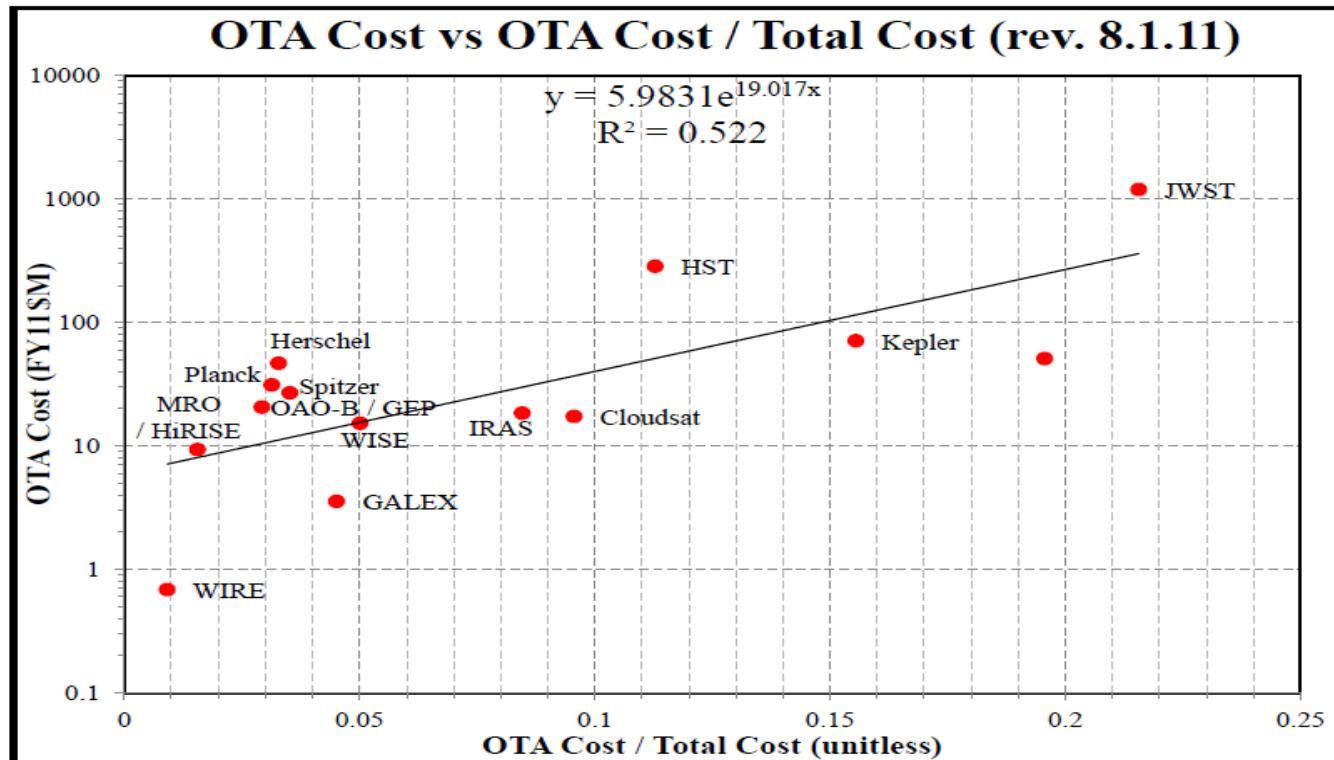


# OTA Cost as a % of Total Mission Cost

OTA Cost varies from approximately 1% to 25% of the Total.

OTA's cost as % of Total depends upon need to develop custom tooling or infrastructure – or use existing.

WIRE is clearly questionable & under review. Also, have asked GALEX to clarify their CADRe cost (missing Structure cost)





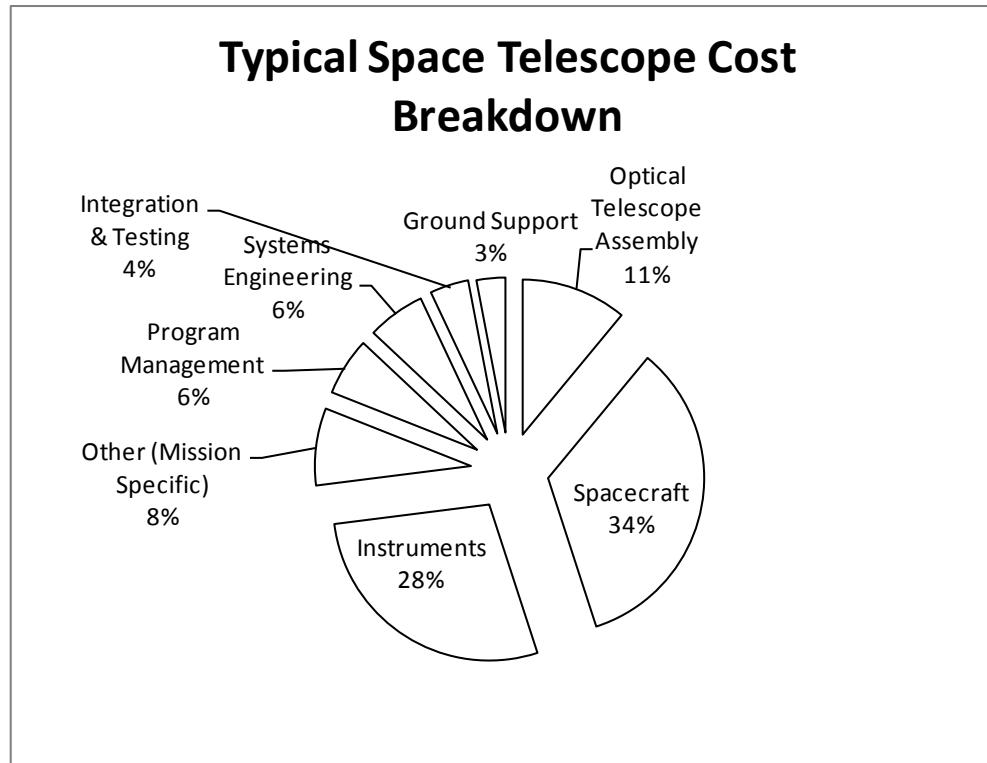
# OTA Cost as a % of Total Mission Cost

We have detailed WBS data for 7 of the 14 free flying missions.

Mapping (5.3.11) database on common WBS gives OTA ~10% of Total

Some say that Power System is 20% of total mission Cost and Mass

For 1960/1970 mission, electronics costs are greater than OTA costs.





# Agenda

- Introduction and Summary
- Data Collection Methodology
- Statistical Analysis Methodology
- What to Model?: OTA or Total Mission Cost
- **Single Variable Modes: Mass and Diameter**
- Multi-Variable Models
- Conclusions



# OTA Cost Regression

Regressing on 15 normal incidence, ‘free-flying’ UVOIR OTAs

Significant Variables: Diameter, Focal Length, Volume, Pointing & Mass

FL has the highest  $R^2_{adj}$  and Mass has the lowest SPE

Volume, FL & Diameter have acceptable  $R^2_{adj}$  & SPE (but they all Dia)

rev. 8.1.11		OTA Cost vs V1													
Variable Name		Aperture Diameter		PM F Len.		PM f/#		OTA Volume		FOV		Pointing Stability		OTA Mass	
Var.	p-value	1.42	0.00	1.55	0.00	0.58	0.57	0.58	0.00	-0.12	0.69	-0.76	0.02	1.08	0.00
Adjusted r <sup>2</sup>		81%		94%		-3%		92%		4%		6%		86%	
SPE		123%		92%		707%		80%		400%		242%		58%	
n		15		11		11		11		12		8		13	

Variable Name		OTA Areal Density		Spectral Range minimum		Diff. Lim. $\lambda$		Operating Temp.		Year of Dev. (exp)		Date of Launch (exp)	
Var.	p-value	0.06	0.90	-0.07	0.56	-0.11	0.54	0.04	0.89	0.00	0.91	0.02	0.56
Adjusted r <sup>2</sup>		-8%		-4%		-7%		-8%		-7%		3%	
SPE		810%		830%		787%		979%		1007%		747%	
n		12		15		12		14		14		15	



# Mass Model



## Mass Model

As an optical engineer, my preference is to develop a model based on an optical parameter, i.e. Aperture Diameter.

Aperture Diameter interests ‘users’ of space telescopes because it is directly proportional to sensitivity and resolution.

But, many believe that Mass is the most important CER.

Total system mass determines what vehicle can be used to launch.

Significant engineering costs are expended to keep a given payload inside of its allocated mass budget.

Such as light-weighting mirrors and structure.

Space telescopes are designed to mass

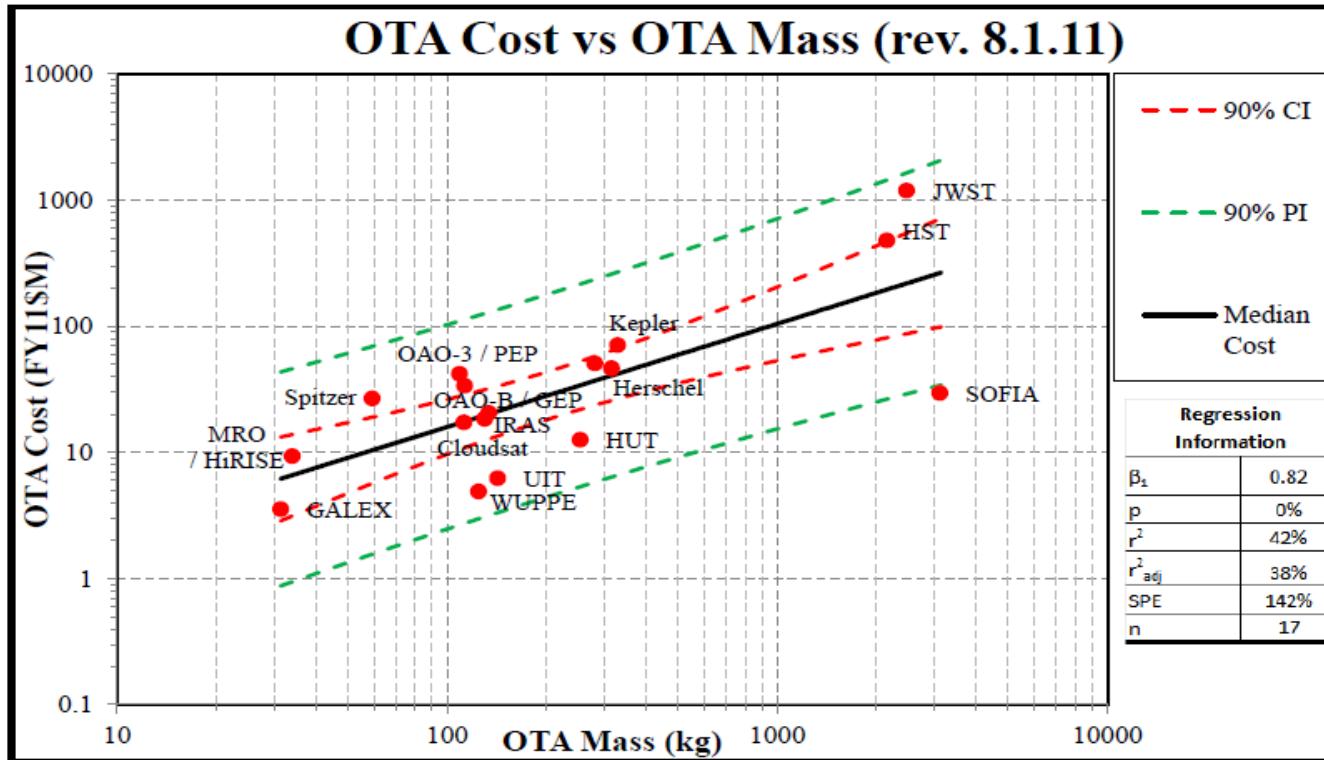


# OTA Cost Mass Model #1

Regressing on all OTAs in the data base:

$$\text{OTA Cost} \sim \text{OTA Mass}^{0.8} \quad (N = 17; r^2 = 42\%; SPE = 142\%)$$

Mass accounts for only 42% of the cost variation & is noisy



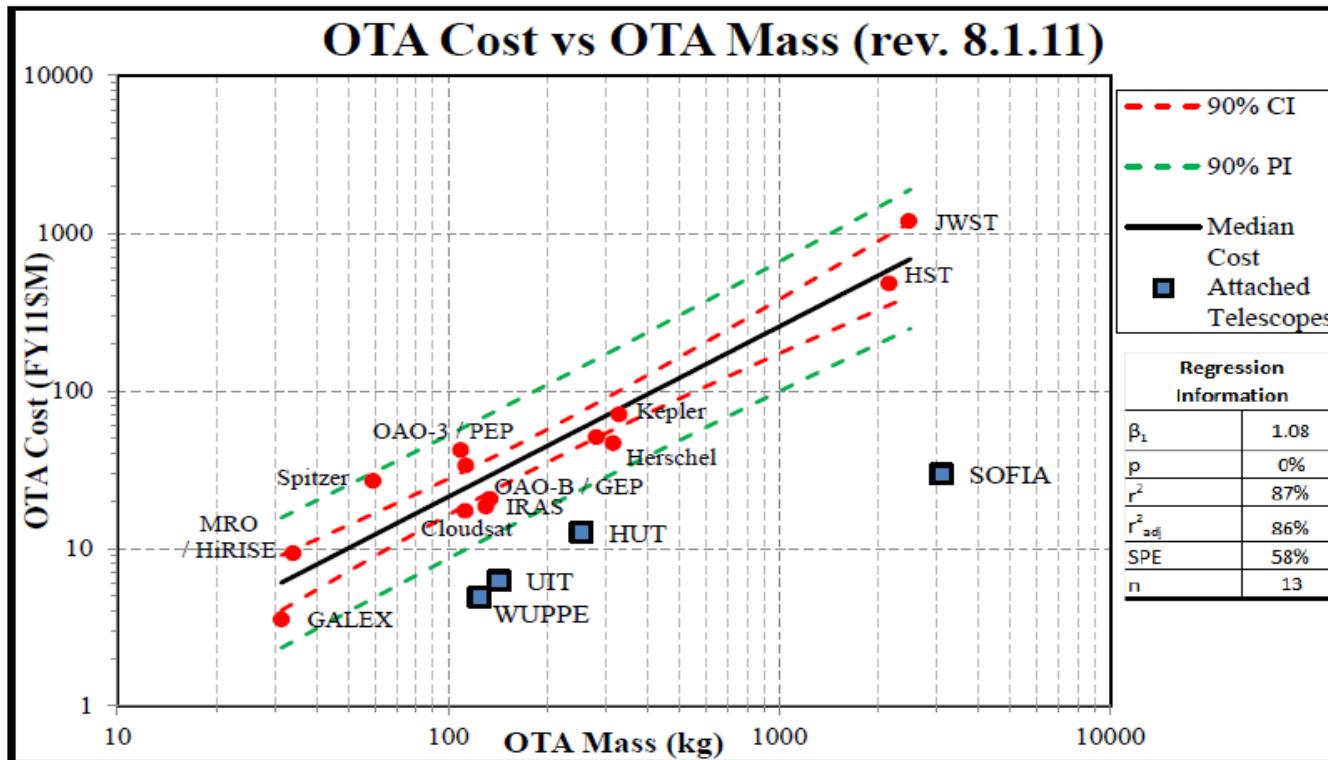


# OTA Cost Mass Model #2

Regressing on only Free-Flyer (excluding ‘attached’ and SOFIA):

$$\text{OTA Cost} \sim \text{OTA Mass}^{1.1} \quad (N = 13; r^2 = 87\%; SPE = 58\%)$$

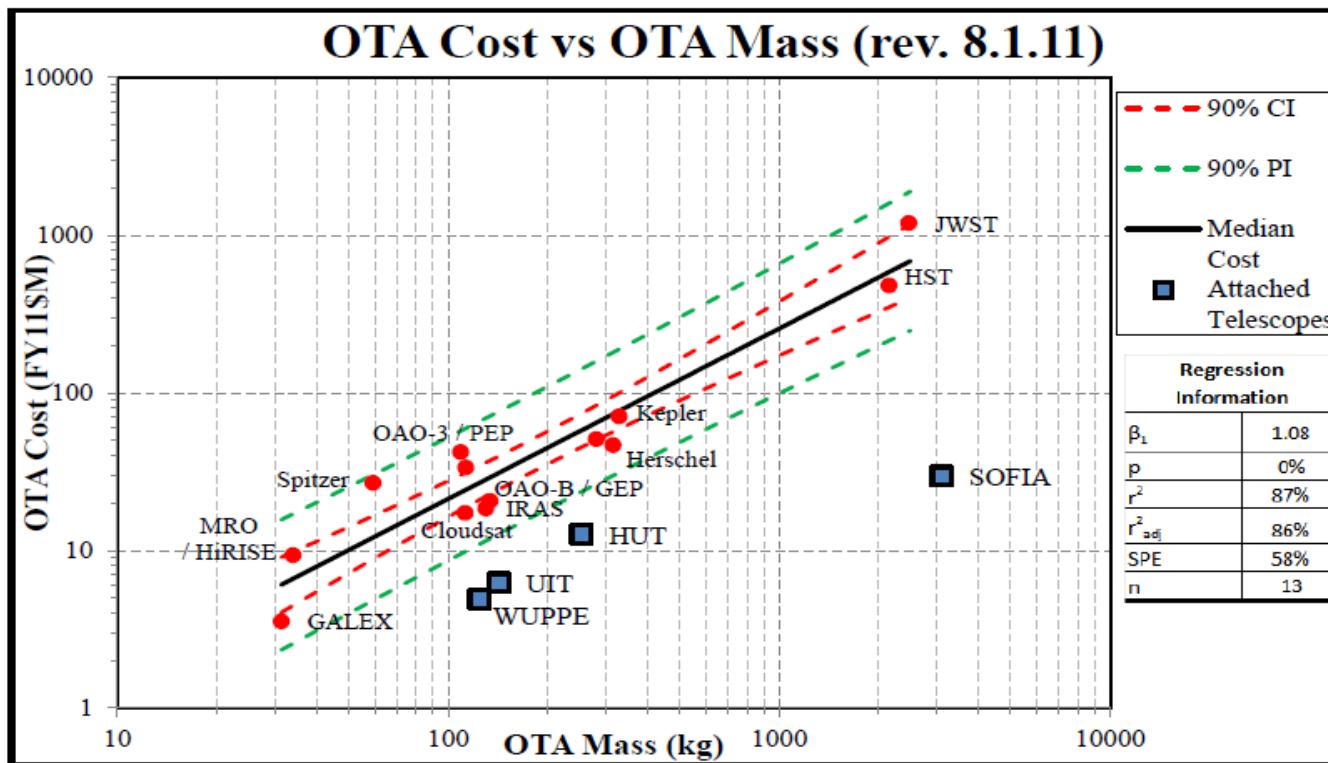
Mass accounts for 87% of the cost variation with less noise.





## OTA Cost Mass Model #2

The 3 ‘attached’ missions & SOFIA clearly are a different ‘class’  
They have a different set of design rules which allow them to  
have a lower cost for a given mass.





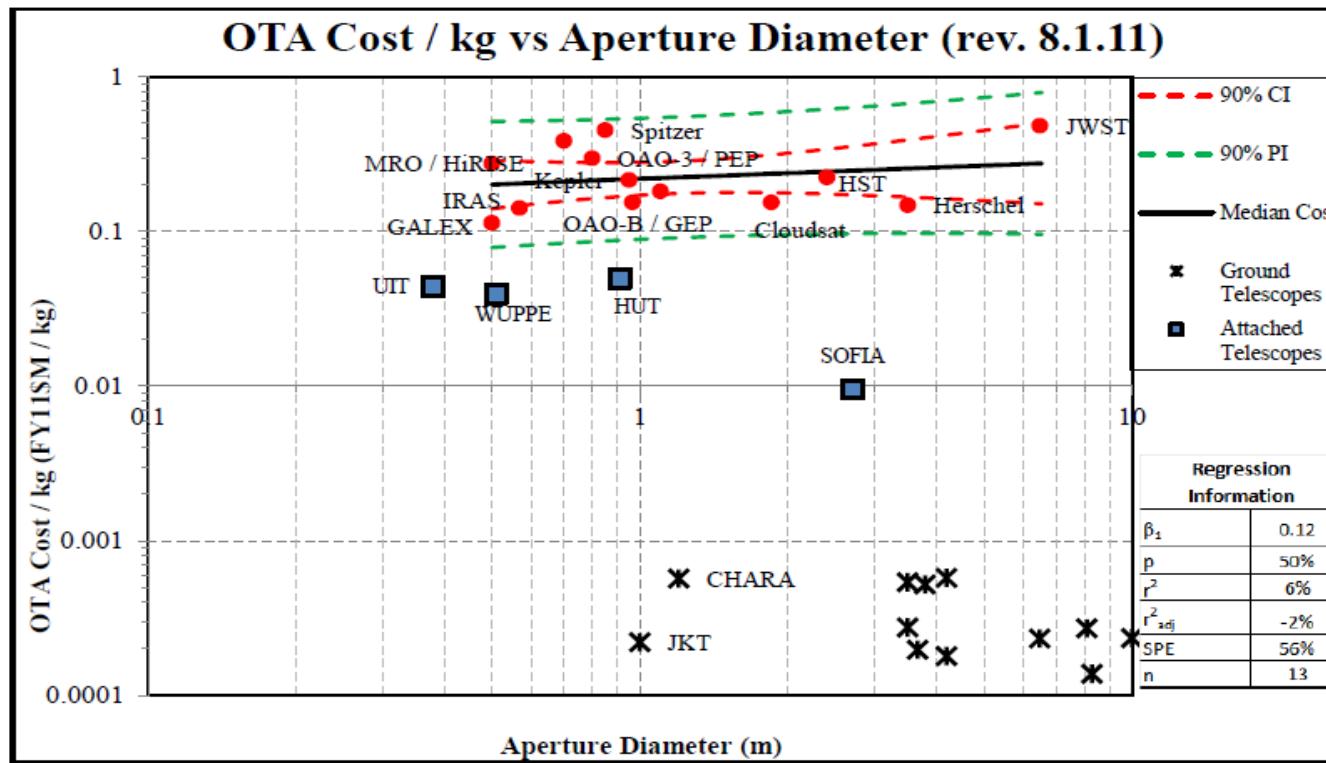
# OTA Cost Density

It costs more to design & build a low mass OTA than a high mass OTA  
Cost per kg depends on mission ‘type’; is independent of aperture size

Free-Flying OTAs are ~2X more expensive per kg than Attached OTAs

Free-Flying OTAs are ~15X more expensive per kg than SOFIA

Free-Flying OTAs are 1000X more expensive per kg than Ground





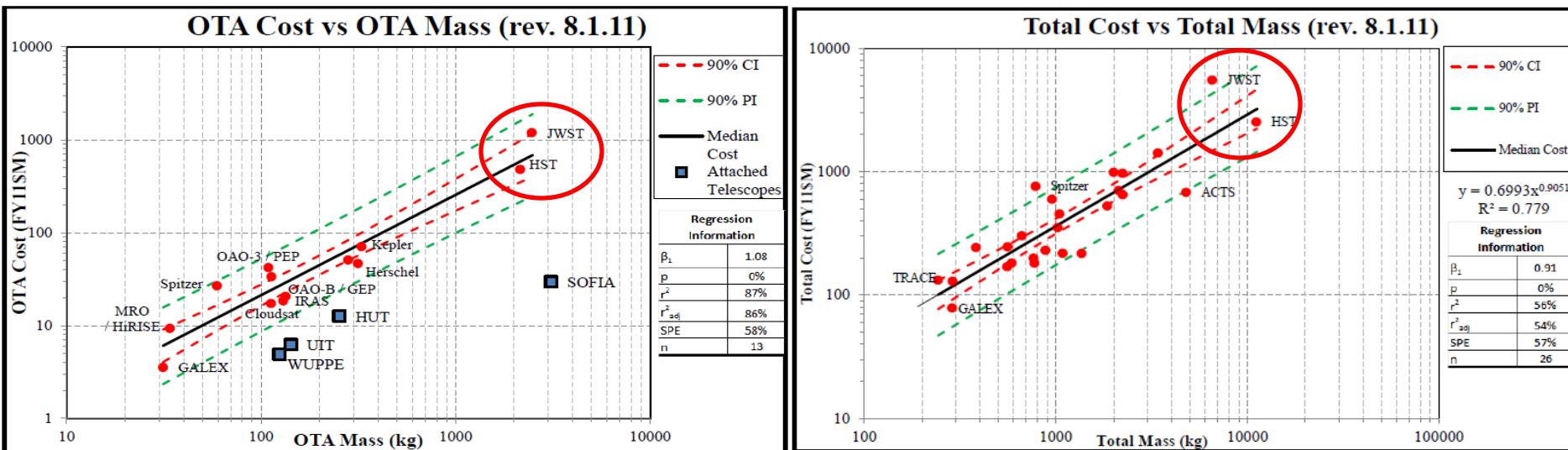
# Mass is not a Good CER

It may appear that Mass is a good CER, but it is not.

JWST & HST have same OTA mass, but JWST OTA costs is 2X HST

HST Total mass is 2X JWST, but JWST Total cost is 2X HST

The reason is complexity – JWST is more complex than HST





## Problem with Mass

Mass may have a high correlation to Cost.

And, Mass may be convenient to quantify.

But, Mass is not an independent variable.

Mass depends upon the size of the telescope.

Bigger telescopes have more mass and Aperture drives size.

And, bigger telescopes typically require bigger spacecraft.

The correlation matrix says that Mass is highly correlated with:

Aperture Diameter, Focal Length and Pointing

But in reality it is all Aperture, the others depend on aperture.



# Aperture Model

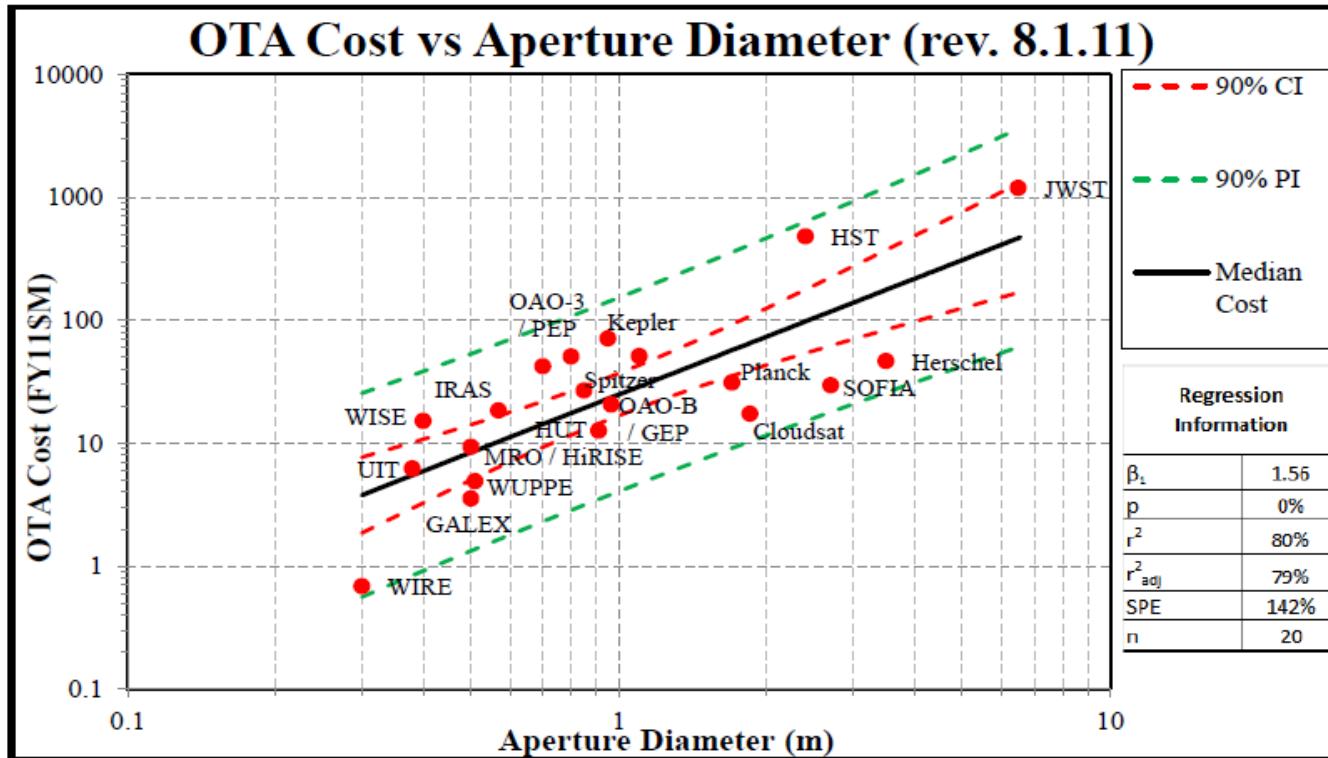


# OTA Cost vs Aperture Model #1

Regressing OTA Cost vs Aperture for all missions in database:

$$\text{OTA Cost} \sim \text{Diameter}^{1.6} \quad (N = 20; r^2 = 80\%; SPE = 142)$$

Diameter accounts for 80% of the cost variation, but is noisy



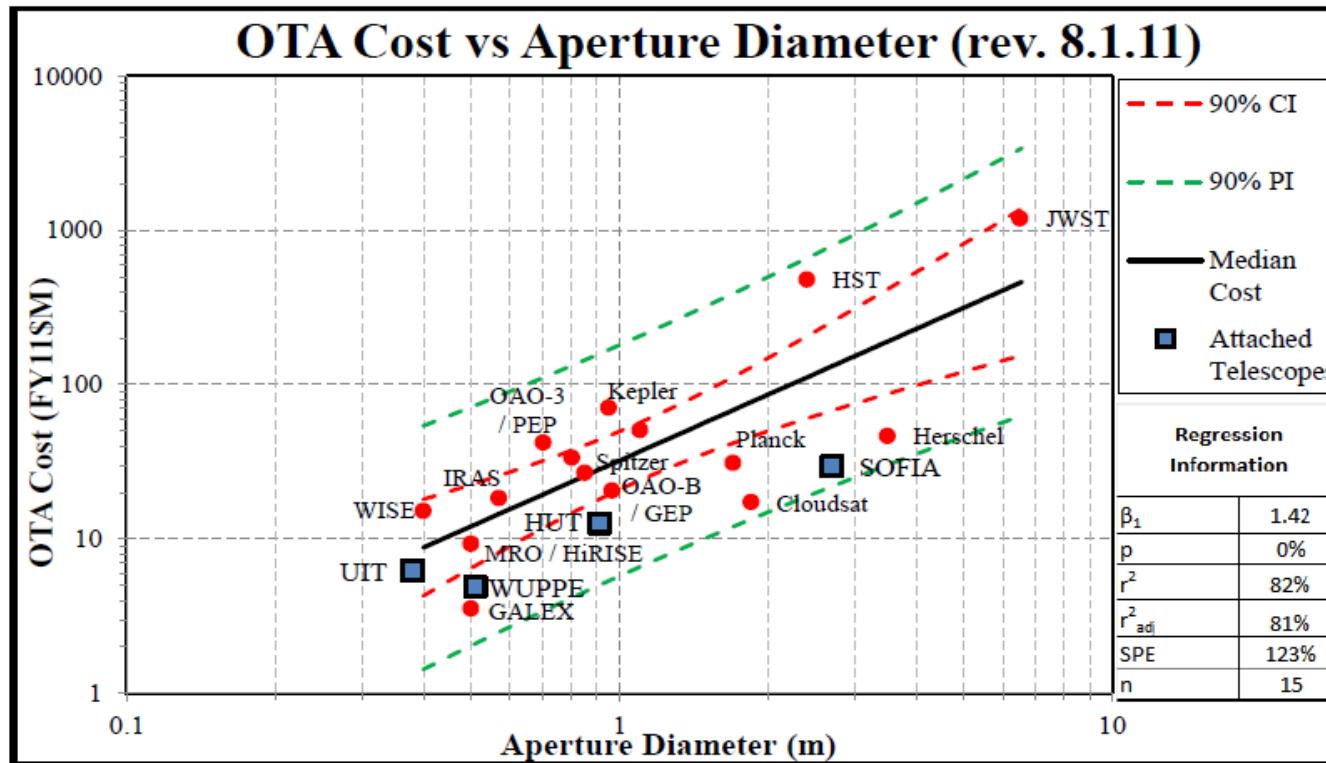


# OTA Cost vs Aperture Model #2

Regressing OTA Cost vs Aperture for just Free-Flyer missions  
(and excluding WIRE):

$$\text{OTA Cost} \sim \text{Diameter}^{1.4} \quad (N = 15; r^2 = 82\%; SPE = 123)$$

Diameter accounts for 82% of the cost variation, is less noisy



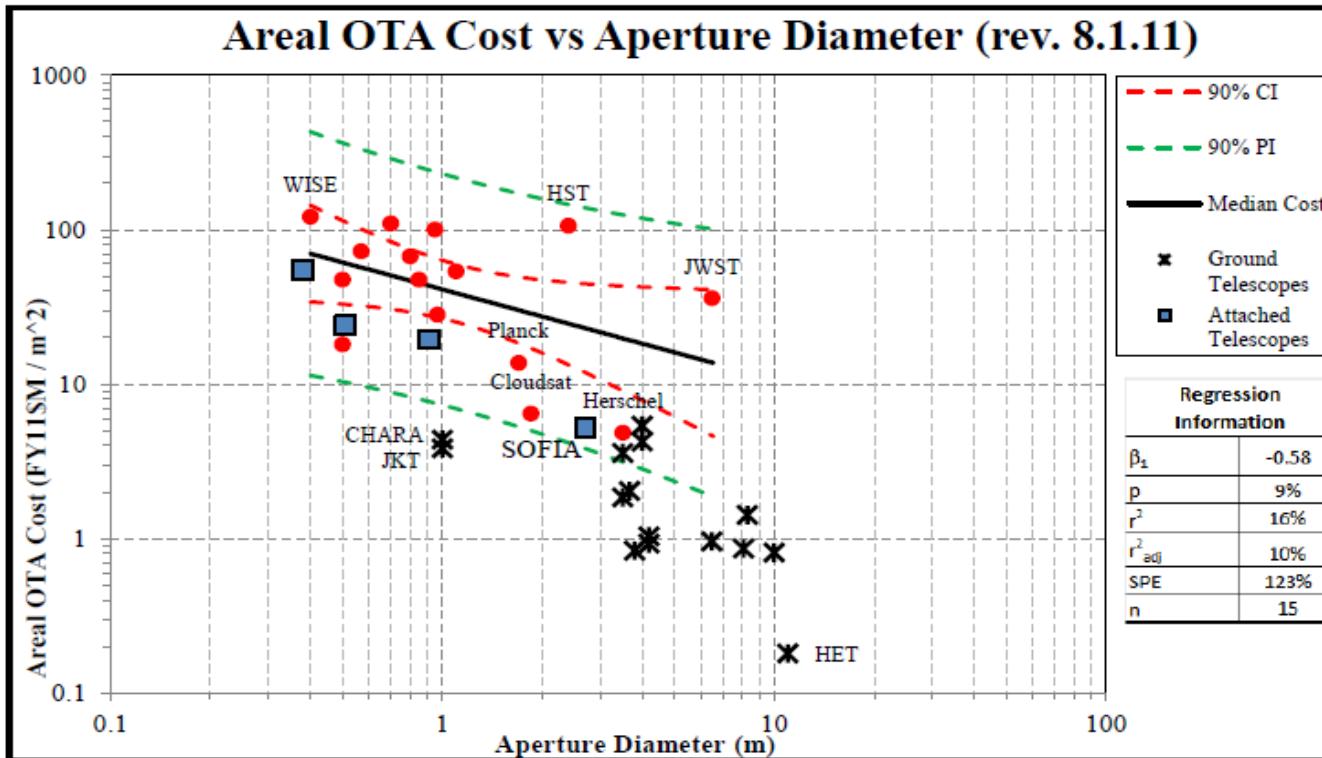


# OTA Areal Cost

Because coefficient for diameter is less than ‘2’, the areal cost (cost per area) decreases as telescopes become larger.

Larger OTAs provide a higher ROI, less \$ per photon.

Also, more massive ‘attached’ and ‘ground’ have lower areal cost





# Agenda

- Introduction and Summary
- Data Collection Methodology
- Statistical Analysis Methodology
- What to Model?: OTA or Total Mission Cost
- Single Variable Modes: Mass and Diameter
- **Multi-Variable Models**
- Conclusions



## Need for a second variable

Assuming that Mass is not the right CER and that Aperture is

Aperture Model only accounts for 70% of the cost variation.

Therefore, other variables must account for the remaining 30% of the cost variation.

Thus, a multi-variable model is required.

First step is a residual analysis.



# How to develop a Multi-Variable Model

Perform multi-variable regression to add a second variable.

Select two variable model based on:

Change in Significance of Diameter to Fit

Significance of Variable #2 to Fit

Increase in  $r^2_{adj}$

Decrease in SPE

Multi-Collinearity

Some variables may increase  $r^2_{adj}$  and/or decrease SPE, but they are not significant or their coefficients are not consistent with engineering judgment or they are multi-collinear.



# OTA Cost versus Diameter and V2

rev. 8.1.10		OTA Cost vs Aperture Diameter and V2													
Variable 2		Aperture Diameter		PM F Len.		OTA Volume		FOV		Pointing Stability		OTA Mass		OTA Areal Density	
Diam.	p-value	1.42	0.00	0.73	0.19	-1.28	0.38	1.26	0.02	1.64	0.01	0.02	0.94	2.05	0.00
Var. 2	p-value	-	-	1.00	0.06	1.00	0.06	0.00	1.00	-0.21	0.32	1.07	0.00	1.01	0.00
Adjusted r <sup>2</sup>		81%		93%		93%		4%		95%		85%		84%	
SPE		123%		84%		84%		142%		66%		58%		54%	
n		15		11		11		12		8		13		12	
Multicollinearity?		N/A		Yes		Yes		No		No		Yes		No	
Variable 2		Spectral Range minimum		Diffraction Limited Wavelength		Operating Temperature		Design Life (exp)		Year of Dev. (exp)		Dev. Period (exp)		Date of Launch (exp)	
Diam.	p-value	1.62	0.00	1.54	0.00	1.49	0.00	0.83	0.02	1.45	0.00	1.14	0.04	1.46	0.00
Var. 2	p-value	-0.18	0.02	-0.22	0.02	-0.08	0.64	0.01	0.01	-0.01	0.46	0.01	0.17	-0.01	0.70
Adjusted r <sup>2</sup>		96%		98%		81%		99%		84%		91%		82%	
SPE		74%		60%		136%		71%		124%		128%		120%	
n		15		12		14		15		14		13		15	
Multicollinearity?		No		No		No		No		No		No		No	

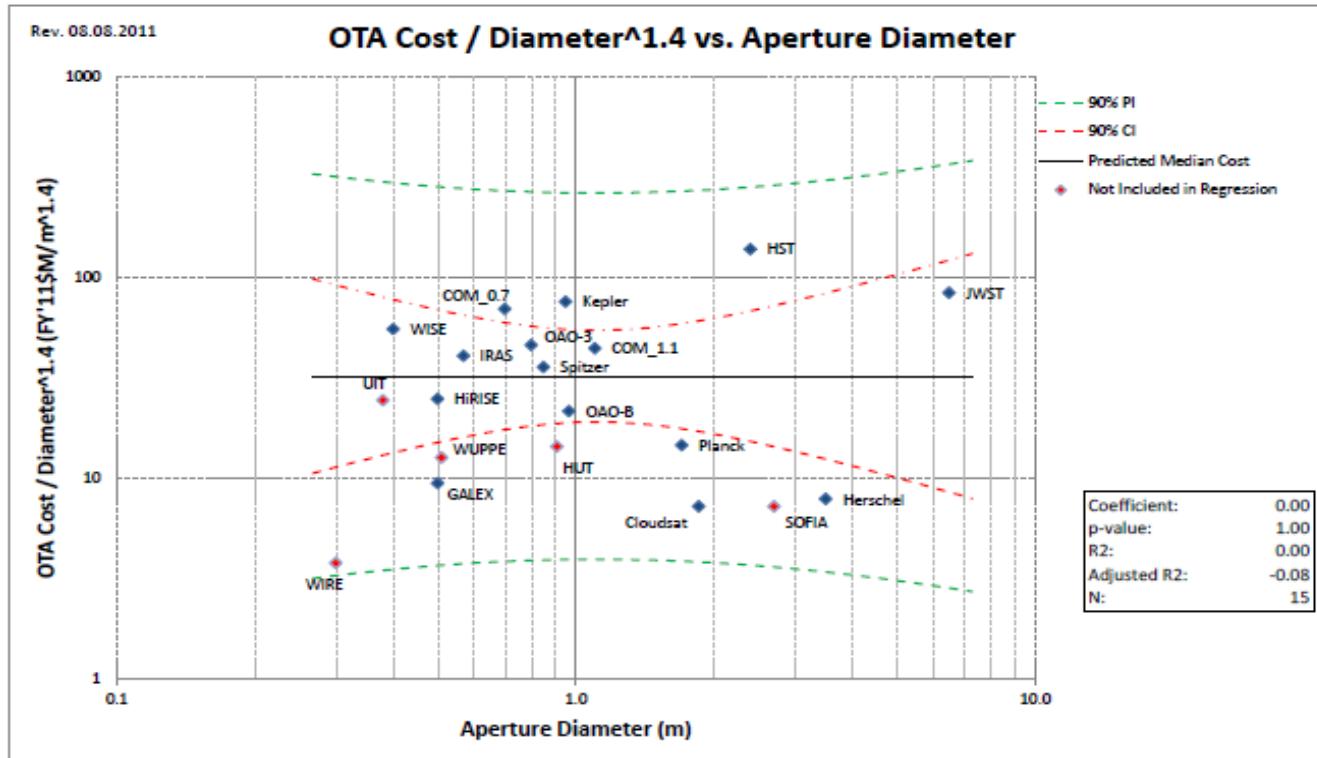
Diffraction Limit & Spectral Min are most significant, both increase R2 & decrease SPE  
 OTA Mass increases R2 to 85%, but is multi-colinear with Aperture Diameter.  
 Other mult-colinear variables are FL and Volume  
 Don't understand impact of Design Life on Diameter.



# Aperture Residual Error Analysis

Divide data by Diameter Model (normalize data) and plot as a function of Variables.

$R^2$  indicates how % of residual error explained by a 2<sup>nd</sup> Variable  
For example, as expected diameter explains ‘zero’ variation

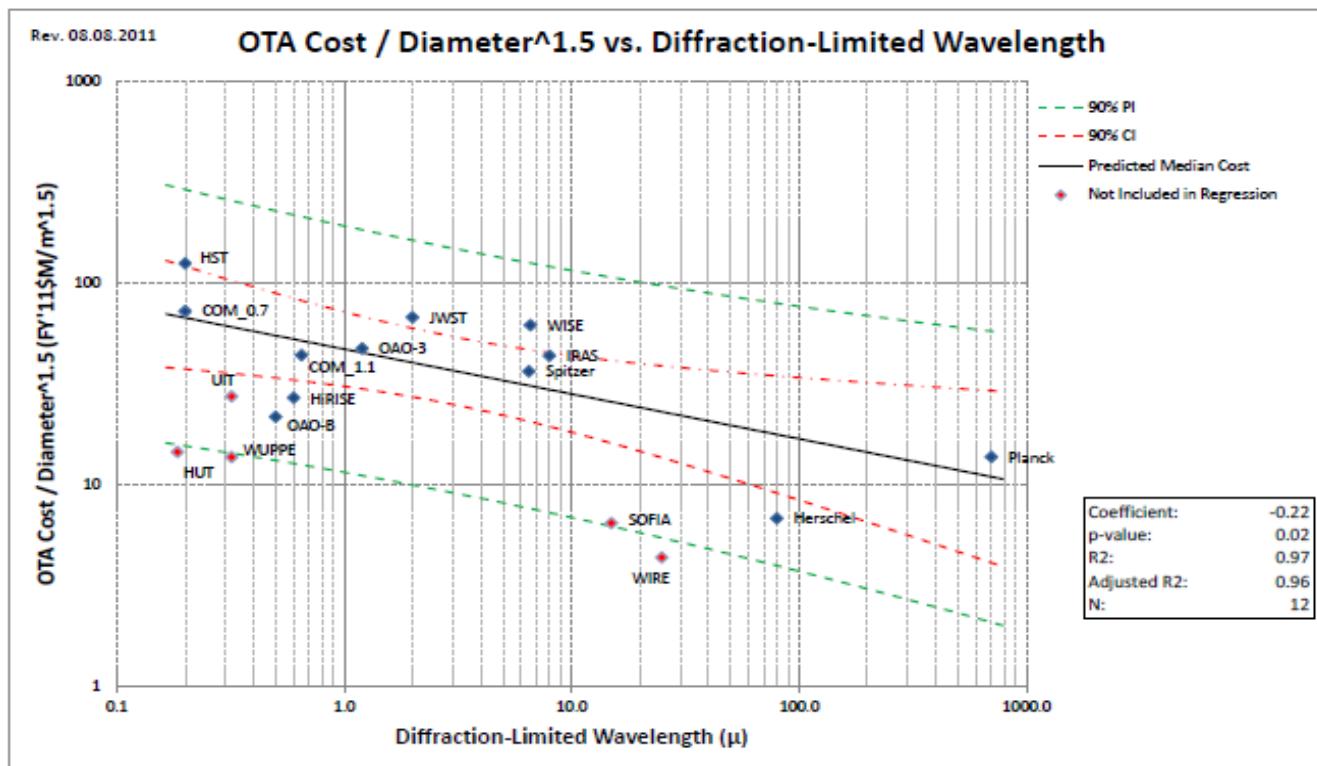




# Aperture Residual Error Analysis: Wavelength

Diffraction Limit Wavelength explains 97% of residual variation

A -0.2 coefficient implies that an OTA with a 10X longer wavelength will cost 40% less.

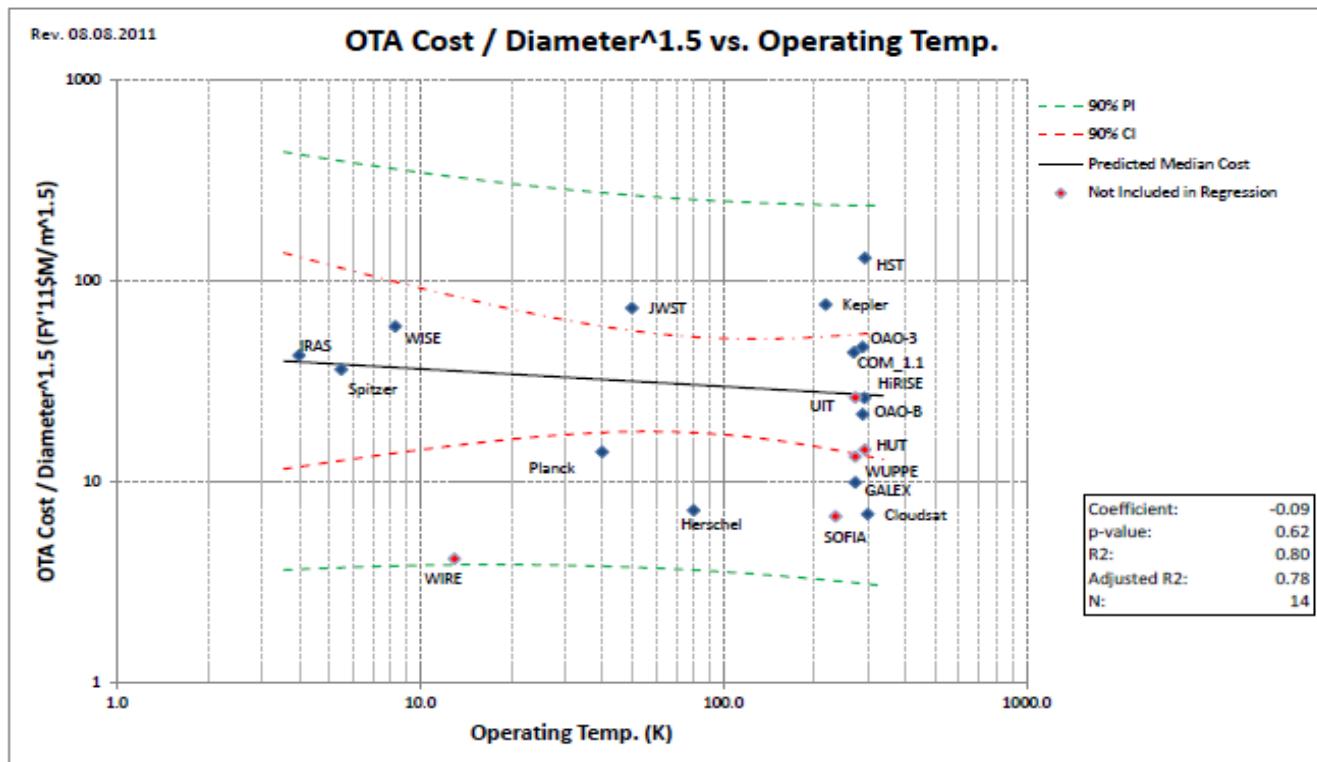




# Aperture Residual Error Analysis: Temperature

Operating Temperature does not significantly explain residual aperture variation

But, it might be a good 3<sup>rd</sup> or 4<sup>th</sup> CER parameter





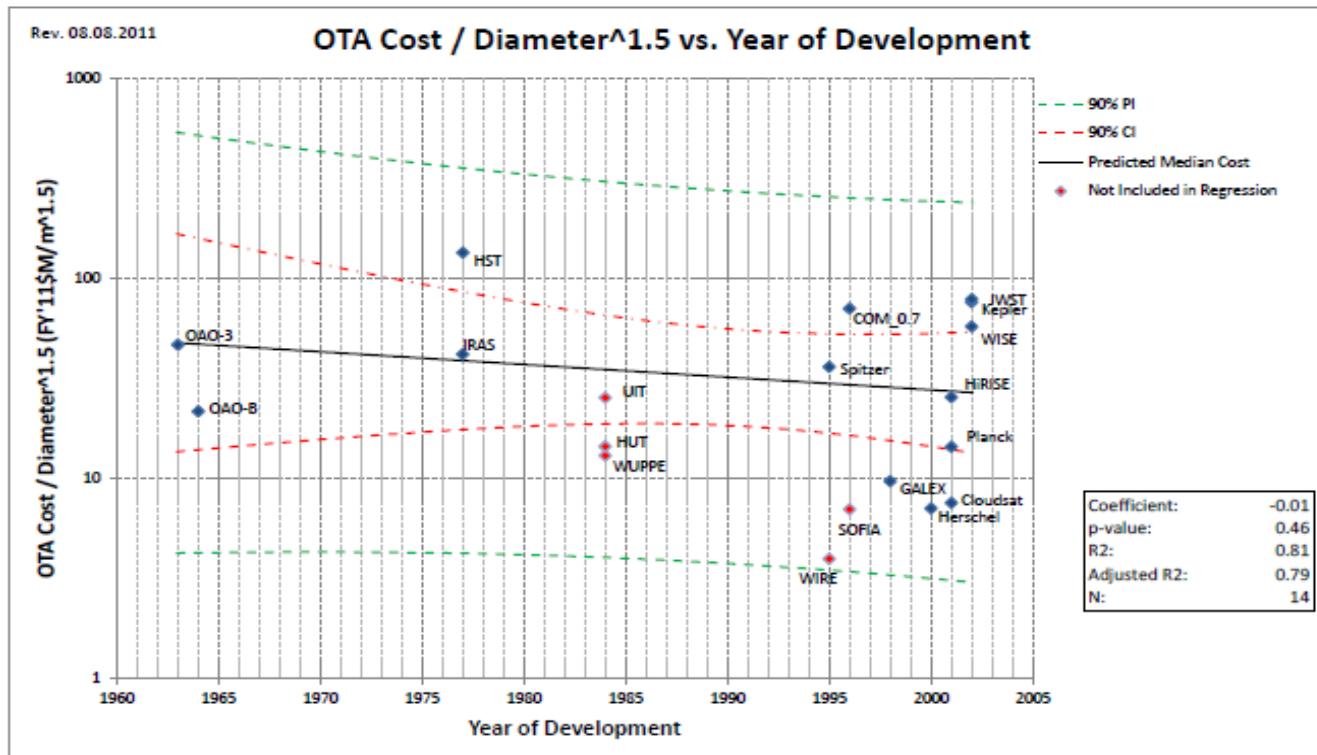
# Aperture Residual Error Analysis: YOD

Year of Development does not significantly explain residual.

But, it might be a good 3<sup>rd</sup> or 4<sup>th</sup> CER parameter

Concern that YOD is correlated with Aperture and Wavelength.

Also, what is role of spectroscopic vs imaging.





## Two Variable Aperture Model

Two second variables best meet all the criteria:

Wavelength Diffraction Limit and  
Spectral Minimum

Diffraction Limited Wavelength yields the best model:

$$\text{OTA Cost} \sim \text{Dia}^{1.6} \lambda^{-0.25} \quad (N = 12, r^2 = 98\%; SPE = 60\%)$$



# OTA Cost versus Diameter, Wavelength and V3

Operating Temperature is the only significant 3<sup>rd</sup> variable

$$\text{OTA Cost} \sim D^{1.7} \lambda^{-0.3} T^{0.25}$$

(N = 11, r<sup>2</sup> = 96%; SPE = 54%)

More effort is required to understand issues related to:

Design Life

Year of Development

rev. 8.1.10		OTA Cost vs Diam., Diff. Lim., and V2									
Variable 2		Diam., Diff. Lim., & V3	FOV		Pointing Stability		OTA Mass		OTA Areal Density		
Diam.	p-value	1.54	0.00	1.37	0.01	1.48	0.10	0.66	0.24	1.97	0.00
Diff. Lim.	p-value	-0.22	0.02	-0.23	0.09	-0.11	0.66	-0.16	0.22	-0.16	0.22
V3	p-value	-	-	0.13	0.66	-0.18	0.56	0.66	0.07	0.66	0.07
Adjusted r <sup>2</sup>		98%		84%		98%		92%		92%	
SPE		60%		73%		49%		46%		46%	
n		12		10		6		10		10	
Multicollinearity?		N/A		No		Yes		Yes		No	

Variable 2		Operating Temperature	Design Life (exp)		Year of Dev. (exp)		Dev. Period (exp)		Date of Launch (exp)		
Diam.	p-value	1.70	0.00	0.87	0.03	1.53	0.00	1.45	0.01	1.49	0.00
Diff. Lim.	p-value	-0.32	0.01	-0.05	0.66	-0.24	0.04	-0.18	0.06	-0.24	0.02
V3	p-value	-0.25	0.10	0.01	0.05	0.01	0.75	0.01	0.42	0.01	0.58
Adjusted r <sup>2</sup>		96%		99%		97%		96%		97%	
SPE		54%		43%		60%		48%		58%	
n		11		12		11		10		12	
Multicollinearity?		No		Yes		No		No		No	



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## Conclusions: Aperture

Consistent with Engineering Judgment Aperture Diameter is a good CER for OTA Cost:

$$\text{OTA Cost} \sim \text{Diameter}^{1.4} \quad (N = 15; r^2 = 82\%; SPE = 123)$$

1 variable only explains 82%, thus a 2 variable model is needed

Two variable model using Wavelength Diffraction Limit explains 98% of data variation with a low SPE.

$$\text{OTA Cost} \sim \text{Dia}^{1.6} \lambda^{-0.25} \quad (N = 12, r^2 = 98\%; SPE= 60\%)$$

In all cases, Areal Cost (\$/m<sup>2</sup>) is less for larger telescopes



## Three Variable Aperture Model

Temperature gives a statistically significant 3 Variable model:

$$\text{OTA Cost} \sim D^{1.7} \lambda^{-0.3} T^{-0.25} \quad (N = 11, r^2 = 96\%; SPE = 54\%)$$

More effort is required to understand issues related to:

Design Life

Year of Development



# Comparison with Historical Models

This study has identified a potential 3 variable model

$$\text{OTA Cost} \sim D^{1.7} \lambda^{-0.3} T^{-0.25}$$

Bely Model (corrected):

$$\text{OTA Cost} \sim D^{1.6} \lambda^{-0.18} T^{-0.2} e^{-0.033(YOD - 1960)}$$

Horak Model:

$$\text{OTA Cost} \sim D^{0.7} \lambda^{-0.18} T^{-0.2} e^{-0.033(YOD - 1960)}$$

But Horak had a different data base.



## Conclusions: Mass

OTA mass is not a good CER

OTA mass is multi-collinear with diameter, and  
more massive telescopes actually cost less to make.

For a given aperture diameter,

Free-Flying OTAs are ~2X more expensive per kg than Attached OTAs

Free-Flying OTAs are ~15X more expensive per kg than SOFIA

Free-Flying OTAs are 1000X more expensive per kg than Ground

Bottom line: using Mass as an OTA CER could easily lead one to  
make inappropriate programmatic decisions.



## General Conclusions

Models are only as good as their data bases – we need more data.

Larger OTAs cost more than Smaller, but Larger Diameter OTAs actually cost less per square meter of Collecting Aperture.

Longer Wavelength OTAs cost less than Shorter.

Cryogenic OTAs may cost less than Ambient.

More study is required regarding cost reduction with year.

If all parameters are held constant, adding mass reduces cost & reducing mass increases cost.